

CHECK WIRING HARNESS BEFORE INSTALLATION

Easier/cheaper to check and recheck all harness's and connections than repairing them later!

INSTALLATION MANUAL

FOR GEM P/N 610C-001
and P/N 1200C-001

Models G1, G2, G3, G4 and G9

DOCUMENT No. 070906

**READ THESE INSTRUCTIONS COMPLETELY
BEFORE PROCEEDING WITH INSTALLATION**

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If You Read Nothing Else Read This!

Follow these recommendations to minimize installation related problems.

USE CAUTION when crimping the terminals onto the end of the GEM harness. Test each crimp by tugging on it sharply. It is almost impossible to pull off a properly crimped terminal.

KEEP the GEM harness at least 1 inch away from the ignition harness, P-leads, and alternator wiring.

Some aircraft produce electrical noise, which will cause erratic indications.

These displays are designed for rugged use. However, since they are constructed of computer-type components, standard precautions for electronic equipment are required;

- Do not expose to rain or water.
- Handle with care - DO NOT DROP.
- Ensure that all electrical connections are correct and properly made.
- Observe electro-static discharge precautions.

The GEM can only do its job of performing accurate engine monitoring if the engine and airframe have a good, solid, low-resistance electrical connection between them. Check for this with a careful visual inspection of the aircraft's ground connections. A measurement with an Ohmmeter will reveal a gross fault but a better test is to measure between the engine case and the airframe with a voltmeter with the engine running and the electrical system under normal load. The voltage difference should be very small (millivolts).

Overlooking this important step can lead to a lot of unnecessary troubleshooting later.

Insight Instrument Corp. supports GEM installations with Insight thermocouple probes and harness. GEM installations with probes and harnesses from other manufacturers are not supported.

NOTE!

Do not use a LED flashlight for the display brightness test! Use a flashlight with an incandescent bulb!

GEM Installation Location Cautionary Notice

The G-series GEM graphic engine monitors have EGT, CHT and TIT primary STC. The GEM display must be mounted so as to be clearly visible to the pilot. The GEM display should be mounted within the pilot's subpanel or the center panel and not on the copilots subpanel unless the GEM can be canted towards the pilot.

GEM Configuration

All G-series GEM's require configuration of important instrument parameters relating to the aircraft, engine, fuel system, etc.

G1 GEM's are **NOT** field-configurable so it is essential to supply all the essential aircraft engine data to Insight at the time of placing the GEM sales order.

G2, G3, G4 and G9 GEM models may be configured in the field but if all the necessary data is supplied to Insight at the time of placing the sales order, the GEM can be pre-configured at the factory. Installation will go quicker and easier if Insight does the configuration for you.

Read This Too!

GEM Fuel Totalizer Cautionary Notice

Please note that not all GEM models provide fuel data. The Fuel Remaining display on the GEM is very useful but is not without limitations. Understand first that the factory fuel quantity gauges are the only instruments in the panel that physically measure fuel level. They are still the primary indication of fuel level in the airplane.

The GEM doesn't measure level, but instead measures only fuel flow rate. The GEM totalizes the rate information to account for fuel used. If you know how much fuel you started with and how much you have used you can figure fuel remaining by simple subtraction. The pilot must supply an accurate starting fuel level for this subtraction to produce the correct fuel remaining result. Should the pilot overstate the fuel quantity on board, the GEM will dangerously overstate the fuel remaining and the endurance time as well. The pilot must be careful and diligent when setting the fuel on board. (Cont'd)

When the aircraft is filled prior to takeoff, the number is the total available on board. If partially filled to a known configuration (ex. tips empty) then the total is easy to calculate.

However if tanks are partially filled on top of a poorly known original value - errors will abound. Unaccounted fuel loss from leakage, fuel vent overflow or theft will of course produce erroneous results.

Be careful and the GEM will deliver safe, reliable, and convenient fuel information. But be sure to cross reference the information on the primary fuel quantity gauges. Never trust a single source of fuel information when you have two on board. Fuel exhaustion still ranks highly among accident causes.

Don't let your engine stop until you're parked.

GEM Engine Temperature Limits (Redlines) Cautionary Notice

Most air-cooled aircraft engines have a maximum allowable Cylinder Head Temperature (CHT) limit specified by the engine or aircraft manufacturer. Similarly if the engine is Turbo-charger equipped a maximum allowable Turbo-charger Inlet Temperature (TIT) limit will be specified (may be labeled as EGT). Most GEM's have 460 F and 1650 F internally set as the CHT and TIT limits as these are the most common values, but there are aircraft that require different values. Be sure to follow the instructions in this document to install the correct Insight P/N so that the GEM's Redlines match the aircrafts specified maximum engine temperatures.

Warranty policy

The Insight Instrument Corp.'s Graphic Engine Monitor temperature display unit is warranted against defects in materials and workmanship for two years from date of purchase. Insight Instrument Corp.'s temperature probes are warranted for one year from date of purchase or 100 hours whichever comes first.

Insight will, at its option, repair or replace, without charge, those products that it finds defective. Material re- turned for repair or replacement will be returned prepaid by second day freight. Insight will not be responsible for repairs that result from improper installation, unauthorized maintenance or abuse.

Insight is not liable for consequential damages or any labor costs, either directly or indirectly. No other warranty is expressed or implied.

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INTRODUCTION

This Installation Manual will acquaint you with the installation requirements of the Insight G-series Graphic Engine Monitors (GEM). Please read it carefully and completely before starting. Insight Instrument Corp.'s Graphic Engine Monitors provides simultaneous analog and digital display of engine temperatures for nearly all makes and models of piston-powered aircraft.

GEM Models and Available Features Table:

<i>GEM Model Name</i>	<i>G9</i>	<i>G4-Single</i>	<i>G4-Twin</i>	<i>G3</i>	<i>G2</i>	<i>G1</i>
<i>Bezel Size</i>	<i>3.125</i>	<i>3.125</i>	<i>3.125</i>	<i>2.25</i>	<i>2.25</i>	<i>2.25</i>
<i>EGT</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>
<i>CHT</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>
<i>TIT</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>
<i>FUELFLOW</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	
<i>OAT</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	
<i>BUSS VOLTAGE</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	
<i>TACHOMETER</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		
<i>MANIFOLD PRESSURE</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		
<i>OIL TEMPERATURE</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		
<i>OIL PRESSURE</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		
<i>ACCELEROMETER</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		
<i>CARBURETOR TEMP.</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>
<i>Data Logging</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	
<i>Probe Diagnostics</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>
<i>RS-232 Comm. to GPS</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		
<i>VIBRATION</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>	<i>YES</i>		

Installation Planning

Plan the location of the GEM display such that it is not positioned near any source of high electrical current and is easily visible by the pilot. If the GEM is to be used as a primary engine instrument it is essential to locate it high in the instrument panel and within the pilot's normal instrument scan.

The GEM's wiring harnesses should be positioned away from sources of high energy, such as ignition harnesses, P-leads, alternator wiring and high frequency radio wiring and antennas.

Some thought about the steps needed may save hours of installation time and trouble-shooting.

Steps to a successful installation

1. Check the GEM's STC Approved Model List for aircraft model eligibility.
2. Check the engines temperature limits (CHT/TIT) as specified in the engine Type Certificate, aircraft Type Certificate or Flight Manual. GEM's are normally supplied with 460 degrees (CHT) and 1650 degrees (TIT) Fahrenheit redlines but others are available. GEM instruments with other redlines are indicated by a suffix in brackets after the instrument Part Number.

Eg. P/N 610C-001 - indicates normal Redlines

P/N 610C-001(X) - indicates other Redlines

Consult with Insight for available temperature limits (Redlines). CHT, EGT, and TIT limit data must be included at the time of ordering a G-Series GEM. The Redline limits are NOT field-configurable.

3. Select which new G-series GEM model is to be installed.

P/N 610C-001 GEMs are configured as models G1, G2, or G3. P/N 1200C-001 GEMs are configured as G4-Single, G4-Twin, or G9. See the "GEM Features" on preceding page.

High-performance aircraft strongly benefit from the advanced features of the G3, G4 or G9 models. The G1 model is designed for small aircraft with low-horsepower engines, it is NOT recommended for complex or high-performance aircraft.

The G9 model is a configuration of the 1200C-001 suitable for radial or liquid-cooled type engines.

- a) Single engine aircraft may have any of G1, G2, G3, G4 or G9 model(s) installed.
 - b) Twin engine aircraft normally have a pair of G3, G4-Single or G9 models installed where sufficient instrument panel space is available, or a G4-Twin instrument may be installed otherwise.
4. Review the features and requirements of the selected GEM model to ensure compatibility with existing and planned aircraft equipment.
 5. Identify whether or not the installation is an upgrade from a legacy GEM system or a first-time GEM installation. In the case of an upgrade, see the "Upgrade Installation" section.
 6. GEM installation kits contain much of what is required for installation except for the materials listed in the "Tools and Materials Required" section. The installer must supply all the materials listed in the "Tools and Materials Required" section.

Upgrade Installations

GEM upgrade kits are designed to utilize existing Insight Instrument Corp. EGT/CHT/TIT/OAT wiring and probes that are already installed in the aircraft. Prior to ordering an upgrade kit, inspect the aircraft's existing GEM wiring and probes for condition, suitability, and length for use with the new GEM. If an existing harness is not suitable for upgrade use, new replacement harnesses are available from Insight.

Legacy GEMs were typically installed as supplemental (non-primary) EGT/CHT/TIT instruments. New G-series GEMs may be used as primary EGT/CHT/TIT instruments. Check the GEM STC for eligibility.

Check the existing GEM instrument-panel location for compliance with the new GEM's location requirements. Primary instruments require clear and easy visibility for the pilot which may not have been required for existing non-primary instrument installations.

Upgrading from legacy GEM models

Upgrading from GEM 602 or 603

Many aircraft have been equipped with GEM 602 or GEM 603 systems. Model 602 systems supported four or six cylinder engines, and 603 systems supported four or six cylinder engines with turbo-chargers. Any 602 or 603 equipped aircraft can be easily upgraded to a G-series GEM system.

New G-series GEMs are plug-compatible with legacy wiring and EGT/CHT/TIT probes from a GEM 602 or 603 system with one difference, the instrument ground wire (black) may have been attached to the engine case and should be moved to an airframe or avionics ground. See wiring drawing 1200-018. The new GEM housings are shorter than old GEM housings, check that the harness length is sufficient to reach the new GEM housing.

The G1, G2 and G3 models have a top control knob or push-button that requires a clearance hole in the instrument panel, check GEM display drawing 610C-001 and outline drawing #930309 for hole sizes and locations.

The G1 GEM is a functional direct replacement for a 602 or 603. Specify four or six cylinder, injected or carbureted, and turbo-charged or non-turbo at time of ordering. The G1 GEM is not field-configurable and must be returned to Insight if a configuration change is needed.

The G2, G3, G4 and G9 models have an additional interface connector (P2) to provide connections to supplemental features such as fuel flow sensors, tachometer ,etc. Upgrade kits contain much of the wiring and sensors to accomplish installation, contact Insight for details. See the following sections of this manual for installation details of the new sensors and wiring.

The G4 models are available in single engine and twin configurations. The G4 displays fit into a standard 3.125 inch panel hole. See drawing #930309.

The G9 model is a single engine GEM that can be configured for radial or liquid-cooled engines.

Upgrading from GEM 610 or GEMINI 1200

Upgrading from models 610 or 1200 is similar to upgrading from 602 or 603 with a few differences.

Previous 610 installations will already have all holes drilled for panel mounting a G1, G2 or G3. G4 models require a 3.125 inch hole.

Previous installations of 610 and 1200 models may already have an Insight OAT probe and wiring that can be retained and connected to a G-series GEM. See wiring drawing 610C-315.

Upgrading to G4 Single

The single engine G4 display (3.125 inch) is plug compatible with the G3 display (2.25 inch). Refer to 610C-315 and 1200-018 wiring diagrams.

Upgrading to G4 Twin

The twin engine G4 display is plug compatible with the EGT/CHT/TIT wiring harnesses of legacy GEMINI 1200 installations. The G4-Twin display has an additional interface connector (P2) to provide connections to additional sensors such as fuel flow and tachometer, etc. Refer to G4-315 and 1200-018 wiring diagrams.

Upgrading GEM installations in twin-engine aircraft

Twin engine aircraft may be equipped with two single-engine GEM displays or a twin engine GEM display. Check the ground wire configuration if upgrading from older GEMs. Wire the thermocouple harnesses according to wiring diagram #1200-018. Install the supplementary feature wiring according to wiring diagram 610C-315 for 25-pin D-sub connectors and diagram G4-315 for 37-pin D-sub connectors.

Tools and Materials Required (NOT included in GEM kit - Installer supplied)

- Proper wire stripping tools.
- Proper crimping tools, Insight recommends;
M22520/2-01 Crimping tool and M22520/2-08 Positioner or equivalent.
- Insertion/extraction tool for Amp pin #205089-1.
- Trip-free, re-settable Circuit-breaker(1 AMP).
- Hardware for mounting and plumbing of the Oil Pressure Sensor, the plumbing for the Fuel Flow sensor, and the plumbing for the Manifold Pressure port.

Unpacking the Graphic Engine Monitor Display

- Carefully inspect the contents of this package for damage. If damage is found, save all packaging so that a claim can be made against the carrier.
- Inspect the contents of the shipment to ensure that all component parts and materials have been supplied.
- Visually inspect all components for proper identification or damage.
- Immediately report any discrepancies to the Insight Customer Service Department.

INSTALLING THE GRAPHIC ENGINE MONITOR DISPLAY

Models G1, G2 and G3 displays mount in a 2.25 inch diameter panel cutout. A new GEM installation will require four mounting holes and two holes for the control knobs in addition to the 2.25 inch hole. See Drawing #610C-001 for a display outline. Insight can supply an adapter plate (P/N 610C-033) to allow mounting a 2.25 inch GEM in a 3.125 inch hole. Beware of third-party adapter plates that may not provide correctly sized or tapped holes.

The G1, G2 and G3 will fit directly into the holes from a previously installed GEM Model 610. Upgrading from GEM 602 or 603 will require a single new hole for the upper control knob.

G4 and G9 models fit into a 3.125 inch diameter hole.

See Insight drawing 930309 for 2.25 and 3.125 inch panel mounting holes.

Connections to the GEM G-series displays shall be made as follows:

G1, G2, G3, G4-Single in accordance with Drawing No's 610C-315 and 1200-018.

G4-Twin in accordance with Drawings No's G4-315 and 1200-018.

G9 for radial engines in accordance with Drawings No's G4-315, 131001 or 131005.

G9 for liquid-cooled engines in accordance with Drawings No's 131201 and 131202.

Care should be given to provide sufficient depth allowance for harness and cable routing at the rear of the display. **Plan for service loops in all the wiring to ease service or adjustments after the installation.**

NOTE: Check for the possible existence of a "double" ground, particularly if the GEM is a replacement for a previously installed GEM display. The GEM must have only one ground connection.

Tandem cockpit aircraft equipped with a G3, G4, or G9 GEM may have a slaved GEM installed in the second instrument panel. The slaved GEM display will continuously show an exact copy of the main GEM screen (EGT/CHT/TIT bar-graph and digital read-outs). Wire the slaved GEM display as shown in Drawings 610C-315 or G4-315.

Installing The Exhaust Gas Temperature Probes

The Exhaust Gas Temperature (EGT) probe is designed fit into a hole in each exhaust stack and be secured with an integral stainless steel clamp. See Drawing No. 8254.

It is important that each probe is mounted a uniform distance from the exhaust stack flange. For normally aspirated engines, a nominal distance of 2 to 3 inches from the flange is recommended.

For turbo-charged engines, a nominal distance of 4 to 5 inches is recommended.

If the recommended distance is impractical because of obstructions, slip joints, or bends in the exhaust system, position all the probes a uniform distance from the flange as space permits.

It is more important that all probes be positioned at a uniform distance from the flange rather than meeting the preferred dimension. Probe locations closer to the flange may result in slightly higher (inconsequential) temperature indications. (Careful matching of probe position will provide best temperature readings.)

If the probe must be positioned in a slip joint the inner tube must have a clearance hole of at least 1/4" diameter to prevent it from shearing the probe. Be certain to locate all holes to allow straight-in insertion of the probe without bending or stressing the probe tip.

Before drilling, ensure that nothing interferes with the probe, clamp, clamp screw or wire. Center punch and pilot drill each hole in the exhaust stack as per Drawing #8254. Use caution while drilling perpendicular to the stack to prevent an elongated hole. NOTE: Tighten the clamp screw with hand-torqued nut driver only.

A right angle drill extension may be necessary in some locations. The probe will slip into a carefully drilled hole and make a tight seal.

Installing The Cylinder Head Temperature Probes

There are three types of Cylinder Head Temperature (CHT) probes:

- Spring Probe (Part Number 2852)
- Spark Plug Gasket Probe (Part Number 2853)
- Adapter Probes (Part Numbers 2855 Bayonet Adapter, and 2856 Threaded Adapter)

The Spring Probe (Part Number 2852) is equivalent to the "old style" Bayonet Probe and screws into threaded thermo-wells in the cylinder head next to the lower spark plug (on top in some engines).

The Spark Plug Gasket Probe (Part Number 2853) replaces the copper 18 mm diameter spark plug gasket. The

Bayonet Adapter Probe (Part Number 2855) screws directly into the temperature well and replaces the standard bayonet adapter to allow simultaneous utilization of the factory-installed Bayonet Probe.

The Threaded Adapter Probe (Part Number 2856) is similar to the 2855 except it is threaded on the inside to allow simultaneous utilization of the factory-installed Threaded CHT Probe.

CHT Probe Characteristics

The Spring Probe (2852) has a probe bushing with a screwdriver slot to facilitate tightening into place. A drop of anti-seizing lubricant on the threads before installation will ease installation, and subsequent removal. When installing Spring Probes be sure you have correctly identified the standard thermo-well. Some aircraft have fuel primer ports with the same thread size.

Many factory-installed CHT displays utilize a single threaded or bayonet-style resistive type probe that occupies one of the thermo-wells. This probe is not electrically compatible with the thermocouple probes required for the Graphic Engine Monitor.

A Spark Plug Gasket Probe (2853) may be used on this cylinder for the GEM series Display, so as to allow the factory-installed display to remain functional.

The Spark Plug Gasket Probe replaces the standard copper spark plug gasket on one spark plug in the selected cylinder. The spark plug chosen should be the one that provides the best correlation with the other temperature probes. On common downdraft-cooled engines the selected cylinder will be the upper plug. Allow enough wire length to move the probe to the other plug if necessary.

The Spark Plug Gasket must be removed and replaced by the Gasket Probe. Annealing of the Gasket Probe is not required or recommended, and the Gasket Probe Does Not require replacement when the spark plugs are changed.

Alternatively, an Adapter Probe may be used instead of a Gasket Probe. An adapter probe will allow both displays to derive their CHT readings from the same thermo-well. For this reason, use of the Adapter Probe is recommended instead of a Gasket Probe.

If the factory-installed CHT display has a bayonet-style probe, the Insight Bayonet Adapter Probe (2855) may be used instead of the Gasket Probe.

The Bayonet Adapter Probe replaces the original bayonet adapter. It has a screwdriver slot to facilitate tightening in place, and placing a drop of non-seizing lubricant on the threads before installation will ease installation and subsequent removal.

Once the Bayonet Adapter Probe has been installed, the original bayonet probe maybe locked in place with a push and a twist.

If the factory-installed CHT display has a threaded probe located in the thermo-well, the Insight Threaded Adapter Probe (2856) may be used. The Threaded Adapter Probe is installed in the thermo-well between the cylinder head and the original CHT probe.

Installing the Carburetor Temperature Probe

The GEM is supplementary for carburetor temperature. Install the Insight Carburetor Temperature probe (Insight P/N 610C-021) into a 1/4-28 tapped hole in the carburetor. Many carburetors have an existing hole for this purpose but it may be plugged and require drilling or tapping. Consult with the engine or carburetor manufacturer to determine the probe-hole location. Ensure the probe does not interfere with any internal carburetor components.

For G2, G3 and G4 GEMs refer to Drawing 610C-315 or G4-315 for electrical connections. The G1 GEM uses pin numbers "A" and "1" on the P1 Edge-Connector. In the case of upgrading from an older GEM system to a G1 GEM check for the possibility of existing wiring to the "A" or "1" pins on the P1 connector. Remove any existing connections to these pins and only connect the Insight Carburetor Probe. See Wiring Diagram Drawing 1200-018 Rev. E or later.

Installing the Turbine Inlet Temperature Probes

There are several types of Turbine Inlet Temperature (TIT) probes that can be used with the GEM system.

- Clamp-Type TIT Probe (P/N 2871) or Dual TIT Probe (P/N 4871) - is identical to the EGT probe except that the clamp diameter is larger.
- Threaded-Type, (7/16-20) TIT Probe, including a weld-on boss (P/N 2872) or Dual TIT Probe (P/N 4872).
- Threaded-Type, (1/4 NPT) TIT Probe (P/N 2873) or Dual TIT Probe (P/N 4873).

The dual-type probes have two thermocouples, with separate signal outputs to drive two instruments.

For aircraft with a threaded fitting boss already welded to the turbine inlet, the two threaded type probes are used - part number 2872 (7/16"-20 Thread) and part number 2873 (1/4 NPT Thread), as applicable. The 2872 Threaded fitting can be installed on any aircraft when a more permanent installation is desired, or when space for the clamp-type probe is limited.

The Probe Type 2872 Boss fits in a 1/2" inch hole and should be welded by an approved exhaust repair facility. Locate and drill the probe hole as described in Section "INSTALLING THE EXHAUST GAS TEMPERATURE PROBES", and refer to Drawing No. 8254 for the recommended location.

NOTE: If you have questions as to the correct type or location of TIT probes, call Insight Product Support for technical assistance.

InstallingThe Manifold Pressure Tee

The GEM is supplementary for manifold pressure. The GEM's manifold pressure sensor(s) are located internally inside the GEM display. The installer must supply the tubing and a "Tee" fitting to connect the GEM to the existing manifold pressure indicator.

For single-engine aircraft refer to Drawing 610C-315 for the location of the G3 Manifold Pressure port (5/16"-24 #2 Flare fitting) or Drawing G4-315 for the G4-single or G9 GEM (use port #4).

For twin-engine aircraft the G4-twin GEM has two manifold pressure ports, refer to Drawing G4-315, use port #1 (top) for the left engine, and port #4 (bottom) for the right engine.

Installing The Oil Temperature Probe

The GEM is supplementary for oil temperature. Install the Insight Oil Temperature probe into an existing tapped hole in an oil gallery in the engine. Insight can supply Oil Temperature probes in a variety of thread sizes to suite most engine types. Consult with the engine manufacturer to determine the probe-hole location and thread size. Ensure the probe does not interfere with any internal engine components. Refer to Drawing 610C-315 for G3 electrical connections or G4-315 for G4.

Installing The Oil Pressure Sensor

The GEM is supplementary for oil pressure. Install the Insight Oil Pressure Sensor (1/8 NPT) on a length of flex-hose "Tee-ed" into the existing oil pressure sensor. The Installer must supply the "Tee" fitting, flexible hose and mounting hardware. Refer to Drawing 610C-315 for G3 electrical connections or G4-315 for G4.

Installing the Tachometer Wiring for RPM

The GEM is supplementary for Tachometer. Insight provides a shielded cable with included protection-resistor (Insight P/N 610C-026) for connection to a magneto. Refer to Drawing 610C-315 for electrical connections for G3 or G4-315 for G4.

For aircraft equipped with electronic ignition systems with a tachometer signal output, use Insight Tachometer Signal Adapter P/N 610C-027 instead of P/N 610C-026. Wire according to Insight wiring diagram #101123.

Installing the Vibration Sensor

The GEM is supplementary for Vibration Measurement. Insight provides a shielded cable pre-terminated with a mating circular plastic connector (CPC) for connection to an Insight Vibration Sensor (Insight P/N 1000-004V).

The sensor should be mounted to an engine case bolt on the top of the engine as close to the propeller as practical. The sensor should be oriented with the nameplate up and the aircraft symbol pointing forward. The sensor may be oriented otherwise if the preferred mounting is impractical, please consult Insight technical support.

For GEM models with a 25-pin D-sub connector (G3, G4-Single) refer to Drawing 610C-315, page 2 for electrical connections. Mount the sensor on the engine, wire as shown and cut the "Configuration loop".

For GEM models with a 37-pin D-sub connector (G4-Twin, G9) refer to Drawing G4-315, page 2. Twin engine aircraft equipped with a G4-twin GEM require a vibration sensor mounted on each engine. Wire as shown and then cut the "Configuration loop" ONLY on the right engine. For G9 GEM installations install one vibration sensor and cut the "Configuration loop".

Aircraft with 24 Volt (28 Volt) electrical systems require a 750 Ohm resistor (Insight supplied) installed in series with each vibration sensor's power supply wire. Aircraft with 12 Volt (14 Volt) electrical systems do not require a resistor.

Older Vibration Sensors and cables were equipped with a 7-pin CPC. New production sensors and cables are equipped with a 6-pin CPC. The two different connector types are not compatible. A cable adapter P/N G4-809A may be installed to allow connection of a 7-pin cable to a 6-pin Vibration Sensor, or 7-pin connectors may be removed from service and new 6-pin connectors installed, consult Insight for details.

If the Vibration Sensor's Connector assembly is located so as to be subject to rain or other fluids, the connector assembly should be protected with liquid-resistant wrap or tubing suitable for location in the engine compartment.

Installing the Outside Air Temperature Probe

The GEM is supplementary for Outside Air Temperature. OAT probe must be mounted in a location which is in free ambient air. Avoid locating the probe adjacent to exhaust pipes, in a confined area, or on the top of the fuselage. Any location where the probe is exposed to direct sunlight or radiant heat from the ground will cause increased temperature readings above ambient air temperature.

Insight recommends mounting the probe in the engine induction air inlet near the engine induction air filter. Probes located there will read well as soon as the engine is started.

Refer to Drawing No. 1200-016 for the mounting dimensions and appropriate hole size to drill.

Mount the OAT probe in the prepared hole and secure the nylon nut being careful not to over-tighten.

If the OAT probe is installed on a pressurized aircraft where bulkhead penetration is necessary and/or potting of the OAT probe is required, be certain to apply "Proseal" or equivalent potting substance prior to tightening. NOTE: Be careful not to get potting substance on the connector.

Installing the Fuel Flow Sensor (for Aircraft without an existing Flow Sensor)

The GEM is supplementary for fuel flow. Aircraft without existing Fuel Flow (FF) sensors should have one Insight-supplied sensor installed per engine, except for engines equipped with pressure-carburetors, which require two sensors per engine. The GEM model G4-twin can support pressure-carburetors by utilizing an Insight Signal Junction Box (Subtractor) P/N 610C-028-S for each engine. See Insight drawing #130523.

Aircraft equipped with TAS-1000 Air Data Systems may share the FF sensor signals with the GEM. Refer to the following installation drawings and the TAS-1000 Installation Manual Doc No. 1000-402.

Insight manufactures a variety of FF sensors for injected and carbureted engines. See drawing 1000-303 (3 pages) for sensor details and part numbers.

For single-engine aircraft or twins with two GEM instruments, wire the Insight FF sensor in accordance with Insight Drawing #610C-315, page 2.

For G4-Twin installations use Drawing G4-315.

In all installations the Insight FF sensor(s) case must be connected to airframe ground.

The installer must determine the correct type and location for the FF Sensor, and supply the plumbing and fittings to hook up the FF Sensor(s) to the fuel lines.

For all fuel flow sensor installations refer to FAA AC 43.13-1B, Chapter 8 Section 2, 8-30, 8-31, 8-37, 8-38.

Installing a Fuel Flow Signal Adapter (for Aircraft with an existing Flow Sensor)

The GEM is supplementary for fuel flow. Aircraft equipped with existing FF sensors may or may not be compatible with the GEM's Fuel Flow signal inputs.

An existing sensor may be part of a primary FF Instrument or it may be supplementary. If the existing FF sensor is primary it should remain in the aircraft to preserve the primary FF instrumentation.

If the existing FF sensor is not part of a primary FF instrument then it may be desirable to remove the existing FF sensor and install the Insight FF sensor as detailed above for aircraft without existing FF sensors.

If the existing FF sensor is going to be retained, it may be connected to the GEM through the use of an Insight Fuel Flow Signal Adapter (FFSA). The Insight FFSA works with most pulse-signal type FF sensors but compatibility with other manufactures' fuel flow sensors cannot be assured.

To connect a GEM to an existing fuel flow sensor/fuel computer system use Insight Fuel Flow Signal Adapter P/N 610C-025. Use Insight wiring diagram #101116.

In the event that the Insight Fuel Flow Signal Adapter P/N 610C-025 is incompatible with an existing FF gage or FF computer (such as utilizing 5-Volt or lower excitation voltage), an alternative P/N 610C-025A (Sensitive FF Signal Adapter) is available from Insight that features higher electrical input impedance and so presents a lighter load to a weak signal. Use Insight Wiring Diagram #150302.

The GEM's K-factor must match the K-factor of the FF sensor to correctly measure fuel flow. Use the GEM's Configuration Page to set the required K-Factor.

Installing a High-flow-rate Fuel Flow Sensor

For aircraft requiring a high flow rate (greater than 60 GPH) fuel flow sensor consult Insight. Install the fuel flow sensor according to the sensor manufacturer's instructions. For sensors that required 12 Volt excitation in aircraft utilizing 24 Volt electrical buss, use Insight 12V Fuel Flow Adapter P/N 610C-029. Make connections as shown in Insight wiring diagram #130410.

Connections for RS-232 Data Communications (G3, G4 and G9 only)

The GEM is supplementary for aircraft equipped with a compatible GPS system. A G3, G4, or G9 may be connected to the GPS's RS-232 communications ports. The GEM can transmit Fuel data in one of several common formats to the GPS. The GEM can receive navigation data from the GPS. Refer to the installation drawings for wiring details. Refer to the GPS system manufacturer for configuration of the GPS to transmit and receive data.

Twin engine aircraft equipped with two (2) G3 or G4-Single GEM instruments may be equipped with an Insight SJB P/N 610C-028-C (Combiner) to combine the two RS-232 data streams transmitted by the two GEM instruments into one RS-232 data stream. This configuration is compatible with many GPS systems which may only be capable of receiving one RS-232 data stream. See Insight drawing #610C-315 page 3. If the aircraft is equipped with a TAS-1000 Air Data Computer the 610C-028-C Combiner is not required as the TAS-1000 supplies the RS-232 data to the GPS system, see Insight drawing #121210.

Aircraft equipped with a TAS-1000 Air Data Computer may be connected to the GEM to datalog in-flight air and fuel data. The GEM and TAS-1000 must be wired to share the pulse FF signals and RS-232 data. Refer to the installation drawings and the TAS-1000 Installation Manual Doc. No. 1000-402.

Wiring Considerations

The GEM is supplied with a factory-assembled EGT/CHT/(TIT) wiring harness configured for the specified number of cylinders and required length (for each engine). The harness edge connector contains a polarization pin that mates with a slot in the display's printed circuit board. This prevents improper engagement of the connector.

Before installing, confirm that the factory fabricated harness connector matches wiring diagram, Drawing # 1200-018. All the red thermocouple wires should be on one side of the connector.

The GEM circuit boards are supported during shipment by small anti-static shipping restraints. Leave these restraints in place during the installation of the display and remove only prior to inserting the harness edge connectors. The EGT and CHT wire length is not critical and may be trimmed to any length as required to fit each probe. CAUTION: Splicing of the thermocouple wire is not recommended. The thermocouple wire alloys are not compatible with soldering.

Note: Plan your Installation to include a service loop in the GEM wiring harness to allow for future adjustments.

Wires for the supplemental features are supplied pre-terminated at the sensor end, but must be terminated at the instrument end during installation. The D-sub connectors (25 and 37 pin) utilize crimp pins, see the materials section for recommended tooling.

Some of the D-sub connectors ground pins are shared by multiple wires, use a "pig-tail" style connection inside the connector shell to accommodate multiple wires connected to a single pin.

Display Power and Ground Wiring Connections

The GEM displays automatically accommodates both 14 and 28 Volt electrical systems. Connect the "red" power lead to a separate trip-free, re-settable circuit breaker (1 Amp) that receives power from the avionics bus or aircraft bus.

If the aircraft does not include an Avionics Master switch circuit or bus, we recommend that one be installed or a separate switch (user provided) to remove power from the Display unit during engine starts.

Connect the ground wire (black) to a common avionics ground bus (airframe ground).

CAUTION: For upgrade installations (from GEM 602 or 603) where the harness is grounded to the engine case, disconnect the ground from the engine case and connect the ground wire to the ground bus (airframe ground). Do not connect to two ground points.

NOTE: Double check the Display ground connection before applying power. Many aircraft have terminal strips under the instrument panel that will appear to be connected to airframe ground and will even measure to ground with an ohmmeter. The terminal strips may instead be connected to ground terminated loads such as landing lights or gear motors. When these loads are activated the voltage on this supposed ground will rise to full bus voltage (14 or 28V). [Extensive damage may result from improper grounding and is not covered under warranty.]

Refer to Drawing No. 1200-018.

EGT Probe Wiring

The temperature probes must be wired with the correct polarity. The EGT probes connect to the harness wires with the yellow jacket. The probe leads and harness wires are color coded (red and yellow) to facilitate correct polarity. Each wire is marked with the cylinder number.

Slide the wire marker down the wire so it remains with the installation for trouble-shooting. Strip the wires according to Drawing No. 8254 and terminate with the crimp-on terminals (provided).

Verify the quality of each crimp with a “sharp” pull on the wire. The terminal should be almost impossible to pull off when crimped correctly. Harness and probe wire colors should match as in Drawing No. 8254.

NOTE: The ring terminals may be crimped with a “service type” tool, however, AMP part number #47386 is recommended. Be sure to test each crimp by pulling on the wire to ensure it won't come out. The most common installation problems are the result of poor quality termination.

CHT Probe Wiring

The CHT temperature probes must be wired with the correct polarity. The CHT probes connect to the harness wires with the black jacket. The probe leads and harness wires are color coded (red and white) to facilitate correct polarity. Each wire is marked with the cylinder number.

Slide the wire marker down the wire so it remains with the installation for trouble-shooting. Strip the wires according to Drawing No. 8254.

Terminate with the crimp-on ring terminals provided. Verify the quality of each crimp with a “sharp” pull on the wire. The terminal should be almost impossible to pull off when crimped correctly.

Harness and probe wire colors should match according to Drawing No. 8254. Insulate and bundle as discussed below.

Routing the EGT/CHT/TIT Wiring Harness (P1 and P3)

It is essential to match the cylinder numbers on all the probes to display the proper information to the pilot.

The probe/harness connections should be insulated with the high temperature fiberglass sleeves provided and routed away from high temperature areas, e.g. exhaust stacks, turbochargers, etc.

The probe wires must not be tied in with ignition, alternator or cabin heater ignition wires because of potential errors in temperature readings.

All wires should be bundled and tied with nylon wire ties or lacing cord and attached to the airframe to prevent damage from vibration and wind buffeting.

The probe wiring harnesses are made of special alloy wire that must not be substituted or extended with cop- per wire.

Do not cut the harness too short. Leave a service loop to allow for service and maintenance or in case adjustments are required in the harness routing.

The power and ground wires are copper and may be extended if necessary, but **NOT** the thermocouple wires.

When the installation is complete all wires should be secured using wire ties and carefully checked for interference, rubbing or chafing with flight control cables, or other moving parts.

A Ferrite RF filter should be installed surrounding the Thermocouple wires and located close to the GEM. This is supplied by Insight on new harnesses, and are available from Insight to add on to existing installations.

Connecting and Routing the P2 Harnesses

The P2 harness connects to the D-sub connector (either 25 or 37 pins) on the back of the GEM. This harness carries the electrical signals for all the GEM's supplementary functions such as Fuel Flow, Oil Temperature and pressure, RS-232 communications, etc.

Route the wiring harness away from any high temperatures and high energy sources and be sure to leave a service loop to allow adjustments and ease maintenance. Route the thermocouple wires in the P2 harness through the Ferrite RF filter (on the P1 harness) as shown in Drawings No. 8252 and 8253.

Use the proper tools listed in the Tools and Materials section to crimp and insert the pins into the D-sub connector body.

The Feature Table on page six indicates the supplementary features available for each GEM model.

The G1 GEM has no P2 connector, all connections are made through P1.

The G2, G3, and G4-Single GEM's have a 25-pin D-sub connector, refer to Drawing No. 610C-315 (Pages 1-3).

The G4-Twin and G9 GEM has a 37-pin D-connector, refer to Drawing No. G4-315 (Pages 1-3).

For aircraft with existing compatible Fuel Flow sensors, use Insight P/N 610C-025 FF Signal Adapter as shown on Drawing No. 101116 (one adapter per FF Sensor).

For twin engine aircraft equipped with two GEM's (G3 or G4-Single or G9) and a TAS-1000 Air Data Computer, refer to Drawing No. 121210.

Checking The Installation

Verify ALL connections before applying aircraft power. Pin 15 (connector P1) is aircraft ground and pin S is approximately +14V DC or +28V DC (See Drawing No. 1200-018).

When power is initially applied, the GEM display will illuminate to full brightness.

Immediately upon power application the GEM will display a status screen with serial number, software version number, etc., for a few seconds. After the status screen, the GEM reverts to the main GEM bar-graph page. EGT and CHT (and TIT) color-coded bar graph columns indicate their respective cylinder temperatures, and supplementary values appear in cyan.

The GEM Display brightness level automatically adjusts to match the ambient light level. The automatic dimming may be tested in bright ambient light by covering the entire face of the Display with the palm of your hand for several seconds. The Display will dim and then brighten when your hand is removed.

In low ambient light, the auto-dimming feature may be tested by shining a flashlight on the display for several seconds. It changes brightness slowly, in discrete steps, to prevent annoying flicker in response to rapid ambient light level changes.

NOTE!

Do not use a LED flashlight for the brightness test! Use a flashlight with an incandescent bulb!

After the tests described above have been performed, check for possible interference with existing avionics by listening for audio interference on Com, Nav, DME, ADF, etc. Interference is uncommon, however, these characteristics should be tested.

If interference is detected, remove power from the Display unit to check if it is the emitter of the interference. If the GEM series Display is the interference source, re-route the wiring harnesses away from affected equipment. Contact Insight Product Support if needed further assistance.

GEM Configuration Pages

G1 GEM's are **NOT** field-configurable so it is essential to supply all the essential aircraft engine data to Insight at the time of placing the GEM sales order. Failure to provide the correct configuration data will require the G1 GEM to be returned to the factory.

G2, G3, G4 and G9 GEM models may be configured in the field but if all the necessary data is supplied to Insight at the time of placing the sales order, the GEM can be pre-configured at the factory. Installation will go quicker and easier if Insight does the configuration for you.

GEM models G2, G3, G4 and G9 are configured by special pages not accessible in normal operation. Once saved, configuration data is stored in secure non-volatile memory until overwritten by new configuration data.

To access the configuration pages turn off the GEM for five seconds, and then apply electrical power while holding in the Page (PG) knob. The GEM will allow access to the Registration, General Info, Engine, Date/Time, K-factor, and Misc Configuration pages. Note that the configuration pages do allow access to calibration or Temperature Limit (Redline) data which are only accessible to Insight authorized personnel (Factory-only).

Information entered on the Registration page will appear in the header section of each datalog file.

The General Info page displays technical diagnostic data.

Engine configuration data must be set to match fuel system and engine specifications and limits.

Date/Time allows setting of the GEM's clock and calendar for datalogging.

K-Factor allows adjustment for non-standard Fuel Flow Sensor pulse rates.

Misc allows for setting of RS-232 data transmit protocol.

Turn the PG knob to switch between pages, use the Select (SEL) knob to scroll and modify configuration items within each page. Under "Save Config" select "YES" push the SEL knob to save changes on each page. After configuration changes have been saved, turn off electrical power for five seconds and then restart the GEM to resume normal operation. Do not perform configuration changes in flight.

Refer to the G-Series Pilot's Guide available from Insight for more information regarding configuration.

GEM Software Updates

Contact Insight Instrument Corp. for information regarding software updates.

FLIGHT TESTING THE GEM

It is essential that the GEM be flight tested.

On power-up the GEM will briefly display a status page showing Part Number, Serial Number, etc. Check to ensure that the REDLINE numbers displayed on the status page match the Aircraft Engines CHT/EGT/TIT limits as specified in the aircraft Type Certificate or Flight Manual.

The GEM must be observed in all phases of flight for consistency in performance. If a problem is noted during the flight test, discontinue the flight test and trouble-shoot the system. The GEM's datalog system automatically logs every flight.

Should you have any questions or problems during installation or trouble-shooting of the GEM don't hesitate to contact Insight Product Support.

WEIGHT and BALANCE DATA

- 1 - G-series GEM Instrument 0.5 lb
- 6 - Clamp EGT Probes 9 oz.
- 6 - Spring CHT Probes 6 oz.
- 6 - Gasket CHT Probes 4 oz.
- 1 - Adapter CHT Probe 1 oz.
- 1 - TIT Probe 1 oz.
- 1 - Oil Temperature Probe 1 oz.
- 1 - Oil Pressure Sensor 2 oz.
- 1 - Vibration Sensor 1 oz.
- 1 - Fuel Flow Sensor 6 oz.
- 1 - FF Signal Adapter 1 oz.
- 1 - Carburetor Temperature Probe 1 oz.
- 1 - 8 ft Wiring Harness 14 oz.
- 1 - 24ft Wiring Harness 34 oz.

Troubleshooting

The following is a compilation of the common symptoms and causes of problems that may be experienced with the GEM Systems. Close examination of these symptoms should assist in identifying the cause of the problem.

Note: All G-Series GEM's have a probe diagnostic page to check the thermocouple wire and probes integrity. For G2, G3 or G4's turn the Page knob to see the probe analysis page. For G1's push the top button to see the probe analysis page. The Diagnostic page shows a pair of numbers for each probe, the two numbers are the resistance (in Ohms) of each of the probes two wires, measured from the instrument to the probes grounded tip. The following example resistance numbers are approximate and some variation is normal and to be expected depending on the length of the wiring and the age of the wiring and probes.

For an 8 foot wiring harness the probe analysis page resistance should be:

EGT; 8 and 3 Ohms, and the CHT; 1 and 6 Ohms.

For a 24 foot wiring harness the probe analysis page resistance should be:

EGT; 26 and 10 Ohms, and the CHT; 4 and 18 Ohms.

Symptom: The GEM Display does not illuminate.

Cause: No ground return at Pin 15, of the edge connector either from the engine case or from airframe ground or supply Voltage is missing at Pin S, of the P1 edge connector from the circuit breaker or bus.

Symptom: One or more columns will not illuminate.

Cause: Try recycling power to check the Display. If all columns do not illuminate, the display has been damaged or is inoperative. Check the probe diagnostic page.

Symptom: One or more columns will not illuminate.

Cause: Check the probe connections, the display will blank columns with poor EGT and CHT connections. Columns 5 and 6 shouldn't illuminate on a 4 cylinder engine. Check the probe diagnostic page.

Symptom: No EGT in one or more columns.

Cause: Errors in harness wiring. Visually check probe connections and polarity. Check the probe diagnostic page or measure the resistance of the lead wire manually. It is approximately 1 ohm per foot. Take extreme care to not damage the connector terminals with the meter probes.

Symptom: No CHT in one or more columns.

Cause: Errors in harness wiring. Visually check probe connections and polarity. Check the probe diagnostic page or measure the resistance of the lead wire manually. It is approximately 1 ohm per foot. Take extreme care to not damage the connector terminals with the meter probes.

Symptom: No CHT in one or more columns.

Cause: Faulty probe. Visually check probe connections and polarity. Check the probe diagnostic page or measure the resistance of the probe manually. It is approximately one ohm. Take extreme care to not damage the connector terminals with the meter probes.

Symptom: Display goes out during engine start.

Cause: Voltage transient or over voltage condition. The display should not be turned on during engine start. An Avionics Master or separate power switch should be installed to apply power to the display.

Symptom: Display is unstable.

Cause: Noisy or defective magneto or ignition harness. Check operation on left and right magnetos. The temperature should rise slowly and may stabilize slightly or completely on one or the other magneto. Single magneto operation will pinpoint the problem to one ignition harness, unless both are faulty. Verify that the probe wires and GEM series system harness are isolated from the ignition harness. If necessary, repair or replace the faulty ignition harness to eliminate ignition related interference.

Symptom: Display is unstable.

Cause: Magneto ungrounded or defective P-lead. Check ignition harness for proper shielding, grounding, and loose spark plug caps. Check magnetos for proper grounding or evidence of arcing. Disconnect magneto P-leads one at a time. If this eliminates or reduces the problem, replace the P- lead. A faulty ignition harness will typically cause all EGT readings to "dance" up and down. The GEM series Display will detect this type of fault long before standard test methods, thus eliminating the potential of more serious problems.

Symptom: Display is unstable.

Cause: Noisy or defective alternator, defective generator or faulty ground connection. Try operation with alternator or generator off. Alternator related interference may indicate faulty commutator.

Symptom: Display is affected by radio transmissions.

Cause: Proximity of probe and/or display unit to the radio power wiring and away from radios and antenna coax. Check radio rack connector for missing 50 ohm matching device. The 50 ohm matching device is a thick washer-like component part that is installed underneath the connector end cap. The end cap will have to be unsoldered to check for the matching device. This seemingly unimportant component is supplied with all connectors and is required for proper operation of the connector. Utilize shielded-twisted pair for power leads.

Symptom: EGT indication is not uniform.

Cause: The EGT indication for a fuel-injected engine will typically vary a bar or two from perfectly uniform when leaned for cruise. Clean the fuel injection nozzles. Non-uniformity is normal in carbureted engines. All cylinders are measured by the same circuitry. It is almost impossible to not have identical response on all channels of the GEM series.

Symptom: A sudden or gradual reduction in the EGT indication can be symptomatic of several engine faults.

Cause: Exhaust leaks above the probe and poor compression due to bad rings, valves or valve guides. If a probe substitution does not reveal a faulty probe, check for mechanical faults in the engine. Consult the GEM series Pilot's Guide for more detailed engine diagnostic information.

Symptom: All EGT readings too high.

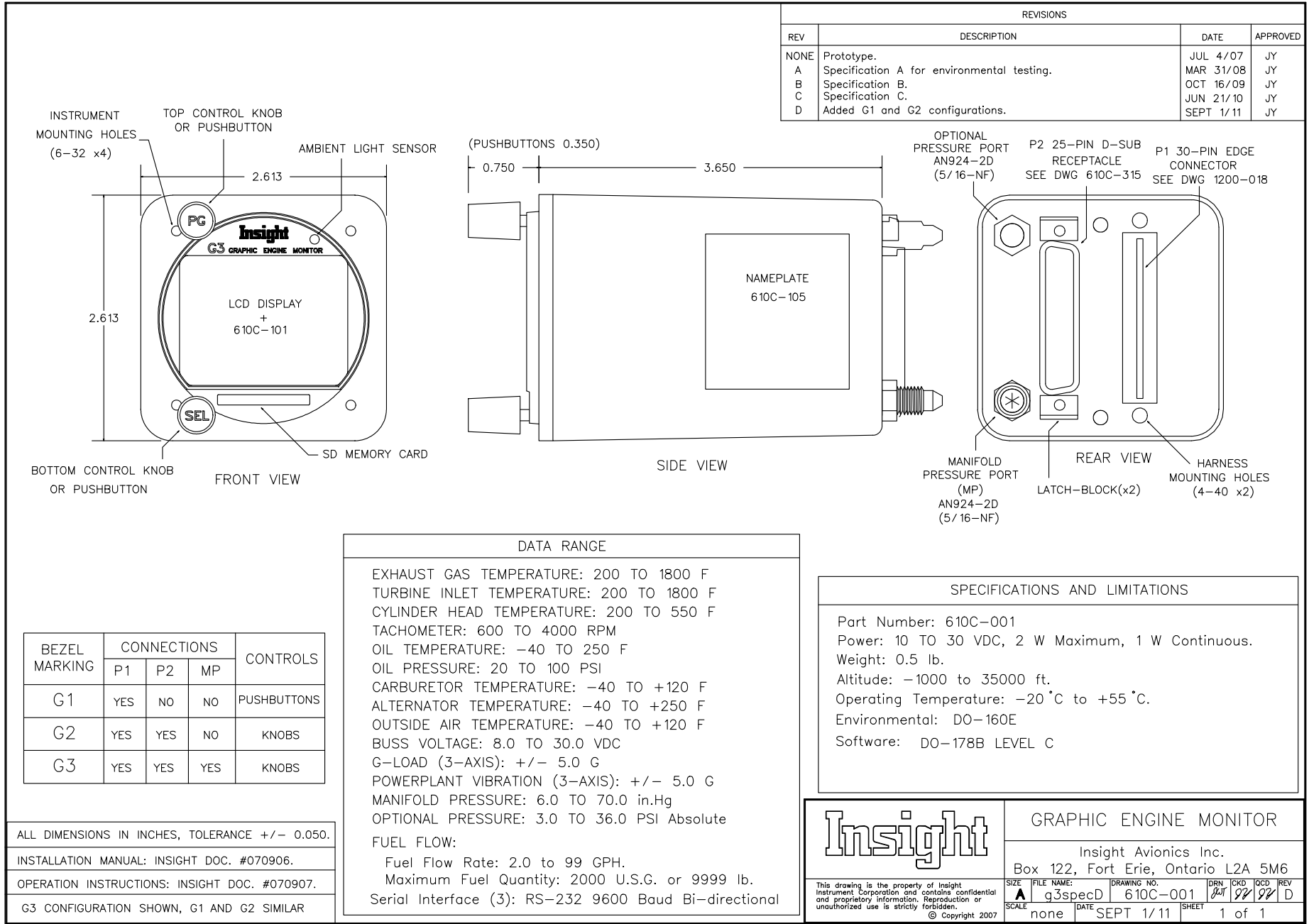
Cause: Readjust the EGT "BAR HEIGHT" on the Display. Use DVM a (digital voltmeter) to measure the difference between the instrument ground and engine block ground with the engine running and the battery charging. If the difference is greater than or 2 V DC with the alternator charging, remove the GEM ground and provide an extension directly to the engine block. If this solution fixes the problem, a ground fault exists between the engine and air-frame which should be remedied.

Symptom: Display indicates full scale or blanks out.

Cause: This can be symptomatic of an intermittent ground fault between engine and air-frame. See "All EGT readings too high".

Drawings

Drawing Number	Rev.	Title	Date
610C-001	D	GRAPHIC ENGINE MONITOR	SEPT 1/11
G4-001	B	G4/G9 GRAPHIC ENGINE MONITOR	JAN 1/14
930309	C	GEM/GEMINI DISPLAY OUTLINES	MAR 31/08
1200-018	E	GEM WIRING DIAGRAM	NOV 1/11
8254	D	GEM PROBE CONNECTION	JAN 1/14
8252	E	GEM CONTINENTAL ENGINES	MAR 31/08
8253	E	GEM LYCOMING ENGINES	MAR 31/08
1200-016	C	OAT PROBE MOUNTING	MAR 31/08
610C-315	F	DSUB-25 PIN INTERCONNECT PAGE 1	JAN 1/14
610C-315	F	DSUB-25 PIN INTERCONNECT PAGE 2	JAN 1/14
610C-315	F	DSUB-25 PIN INTERCONNECT PAGE 3	JAN 1/14
G4-315	C	DSUB-37 PIN INTERCONNECT PAGE 1	OCT 1/13
G4-315	C	DSUB-37 PIN INTERCONNECT PAGE 2	OCT 1/13
G4-315	C	DSUB-37 PIN INTERCONNECT PAGE 3	OCT 1/13
131001	None	G9 WIRING DIAGRAM	OCT 1/13
131005	None	G9 14-CYLINDER WIRING	OCT 1/13
131201	None	G9 EGT&TIT LIQUID COOLED	DEC 1/13
131202	None	G9 DSUB-37 LIQUID COOLED	OCT 1/13
1000-303	E	FUEL FLOW SENSOR Page 1	NOV 30/12
1000-303	E	FUEL FLOW SENSOR Page 2	NOV 30/12
1000-303	E	FUEL FLOW SENSOR Page 3	NOV 30/12
101116	C	FUEL FLOW ADAPTER WIRING	MAR 2/15
150302	None	SENSITIVE FUEL FLOW ADAPTER	MAR 2/15
130410	None	12V FUEL FLOW ADAPTER	APR 10/13
130523	None	FUEL FLOW SUBTRACTOR	APR 10/13
121210	B	GEM / FF/ TAS / GPS INTERCONNECTION	OCT 1/13



REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
NONE	Prototype.	JUL 4/07	JY
A	Specification A for environmental testing.	MAR 31/08	JY
B	Specification B.	OCT 16/09	JY
C	Specification C.	JUN 21/10	JY
D	Added G1 and G2 configurations.	SEPT 1/11	JY

DATA RANGE	
EXHAUST GAS TEMPERATURE:	200 TO 1800 F
TURBINE INLET TEMPERATURE:	200 TO 1800 F
CYLINDER HEAD TEMPERATURE:	200 TO 550 F
TACHOMETER:	600 TO 4000 RPM
OIL TEMPERATURE:	-40 TO 250 F
OIL PRESSURE:	20 TO 100 PSI
CARBURETOR TEMPERATURE:	-40 TO +120 F
ALTERNATOR TEMPERATURE:	-40 TO +250 F
OUTSIDE AIR TEMPERATURE:	-40 TO +120 F
BUSS VOLTAGE:	8.0 TO 30.0 VDC
G-LOAD (3-AXIS):	+/- 5.0 G
POWERPLANT VIBRATION (3-AXIS):	+/- 5.0 G
MANIFOLD PRESSURE:	6.0 TO 70.0 in.Hg
OPTIONAL PRESSURE:	3.0 TO 36.0 PSI Absolute
FUEL FLOW:	
Fuel Flow Rate:	2.0 to 99 GPH.
Maximum Fuel Quantity:	2000 U.S.G. or 9999 lb.
Serial Interface (3):	RS-232 9600 Baud Bi-directional

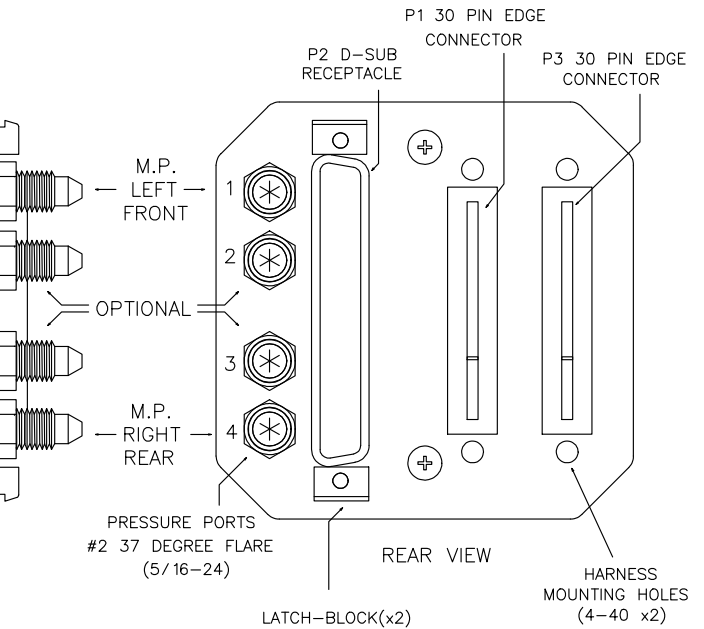
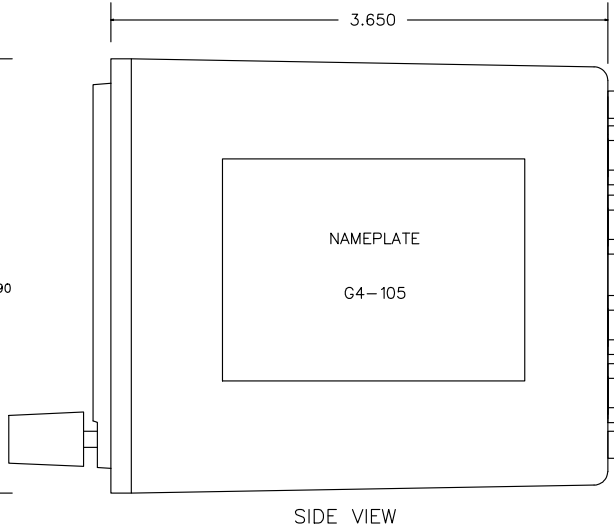
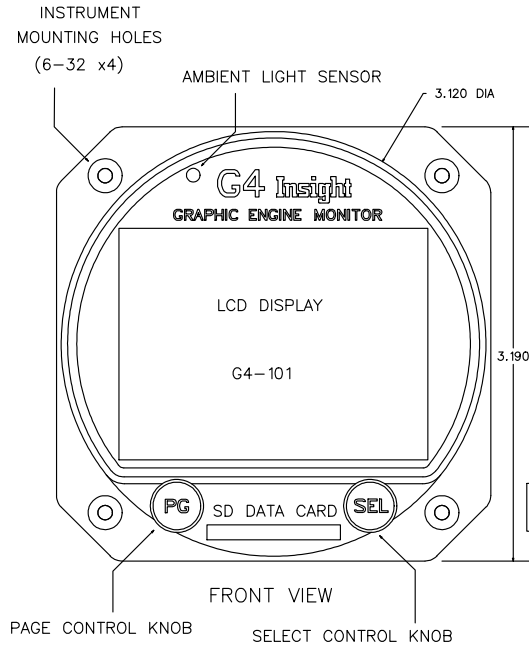
SPECIFICATIONS AND LIMITATIONS
Part Number: 610C-001
Power: 10 TO 30 VDC, 2 W Maximum, 1 W Continuous.
Weight: 0.5 lb.
Altitude: -1000 to 35000 ft.
Operating Temperature: -20 °C to +55 °C.
Environmental: DO-160E
Software: DO-178B LEVEL C

BEZEL MARKING	CONNECTIONS			CONTROLS
	P1	P2	MP	
G1	YES	NO	NO	PUSHBUTTONS
G2	YES	YES	NO	KNOB
G3	YES	YES	YES	KNOB

ALL DIMENSIONS IN INCHES, TOLERANCE +/- 0.050.
INSTALLATION MANUAL: INSIGHT DOC. #070906.
OPERATION INSTRUCTIONS: INSIGHT DOC. #070907.
G3 CONFIGURATION SHOWN, G1 AND G2 SIMILAR

	GRAPHIC ENGINE MONITOR			
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2007	SIZE: A SCALE: none	FILE NAME: g3specD DATE: SEPT 1/11	DRAWING NO.: 610C-001 SHEET: 1 of 1	DRN: [initials] CKD: [initials] OCD: [initials] REV: D

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
NONE	Prototype.	MAR 31/08	JY
A	Added 3rd serial port to P2.	JUN 21/10	JY
B	Added G9 configurations.	JAN 1/14	JY



CONFIGURATIONS				
MODEL	P1	P2	P3	MP
G4 SINGLE	30PIN	25PIN	N/A	1
G4 TWIN	30PIN	37PIN	30PIN	2
G9	30PIN	37PIN	30PIN	1

ALL DIMENSIONS IN INCHES, TOLERANCE +/- 0.050.

MODEL G4 SHOWN, MARK MODEL G9 BEZEL "G9".

DATA RANGES

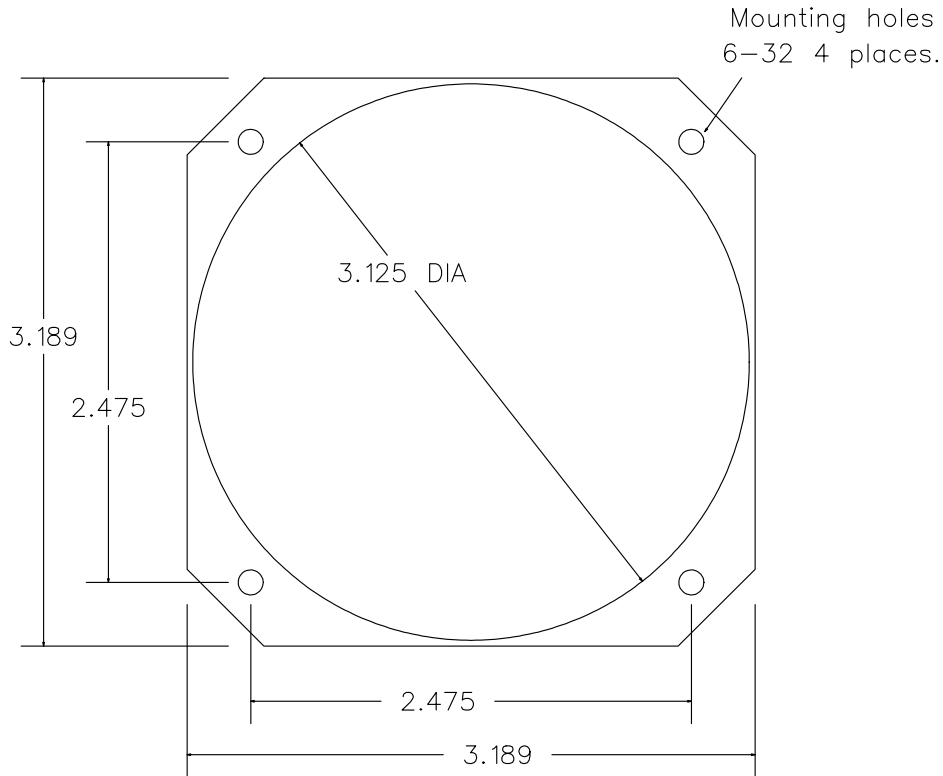
EXHAUST GAS TEMPERATURE: 200 TO 1800 F
 TURBINE INLET TEMPERATURE: 200 TO 1800 F
 CYLINDER HEAD TEMPERATURE: 200 TO 550 F
 TACHOMETER: 700 TO 3400 RPM
 OIL TEMPERATURE: -40 TO +250 F
 OIL PRESSURE: 20 TO 100 PSI
 CARBURETOR TEMPERATURE: -40 TO +120 F
 ALTERNATOR TEMPERATURE: -40 TO +250 F
 OUTSIDE AIR TEMPERATURE: -40 TO +120 F
 COOLANT TEMPERATURE: -40 TO +250 F
 BUSS VOLTAGE: 8.0 TO 35.0 VDC
 G-LOAD (3-AXIS): +/- 5.0 G
 POWERPLANT VIBRATION (3-AXIS): +/- 5.0 G
 FUEL PRESSURE: 0 TO 35 P.S.I.
 M.P. (Standard): 6.0 TO 70.0 in.Hg (35 P.S.I. Max.)
 M.P. (High): 6.0 TO 118 in.Hg (58 P.S.I. Max.)
 OPTIONAL PRESSURE (2): 3.0 TO 36.0 PSI Absolute
 FUEL FLOW:
 Fuel Flow Rate: 2.0 to 999 GPH.
 Maximum Fuel Quantity: 2000 U.S.G. or 9999 lb.
 Serial Interface (3): RS-232 Bi-directional

SPECIFICATIONS AND LIMITATIONS

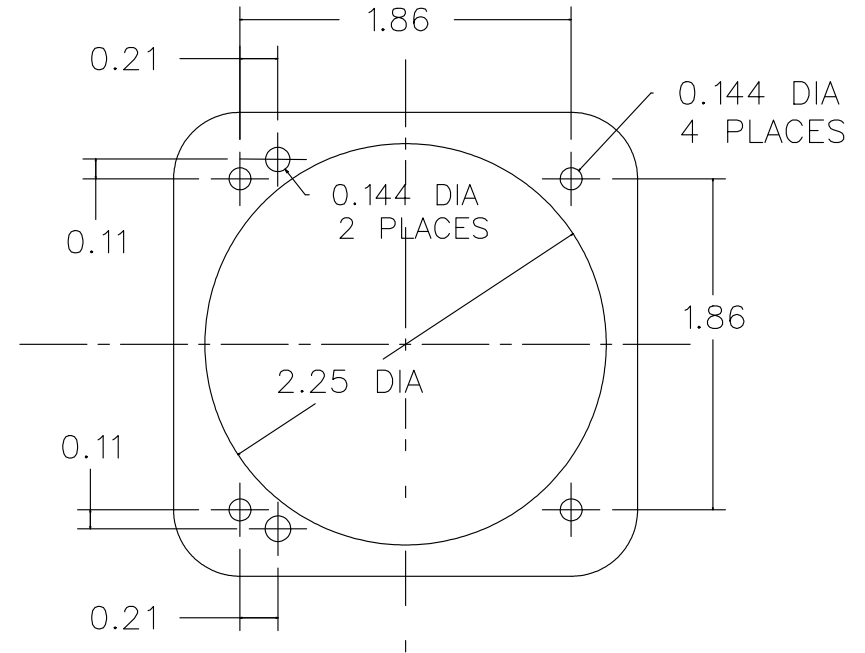
Part Numbers: 1200C-001, 1200C-002
 Power: 10 TO 30 VDC, 2 W Maximum, 1 W Continuous.
 Weight: 0.5 lb.
 Altitude: -1000 to 35000 ft.
 Operating Temperature: -20 °C to +55 °C.
 Environmental: DO-160E
 Software: DO-178B LEVEL C

	G4 GRAPHIC ENGINE MONITOR			
	Insight Instrument Corp. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2007	SIZE: A	FILE NAME: G4specB	DRAWING NO: G4-001	DRN: JAT CHK: JY DCC: JY REV: B
SCALE: none	DATE: JAN 1/14	SHEET: 1 of 1		

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	Addition of detail	930601	JWT
B	CORRECTED HOLE DIMENSION	031114	R.R.
C	REVISED FOR CLARITY	080331	JY



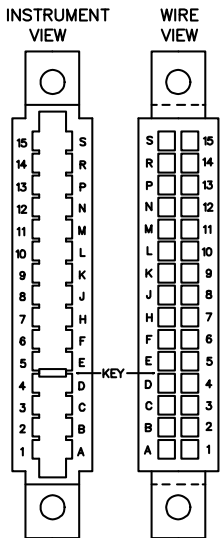
INSTRUMENT PANEL HOLE LAYOUT



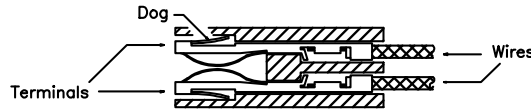
G3 PANEL HOLE LAYOUT
(FOR P/N 610C-001)

<h1>Insight</h1>	GEM/GEMINI DISPLAY OUTLINES					
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6					
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2008	SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD
		gemdoutc	930309	RR	JBK	JWT
SCALE	DATE	SHEET				
NONE	080331	1 OF 1				

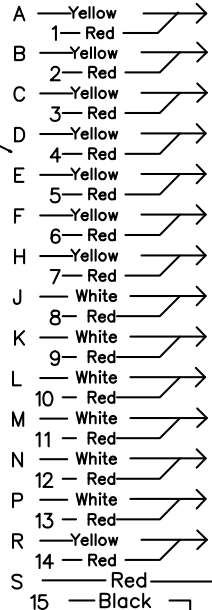
EDGE CONNECTOR



CONNECTOR CROSS-SECTION

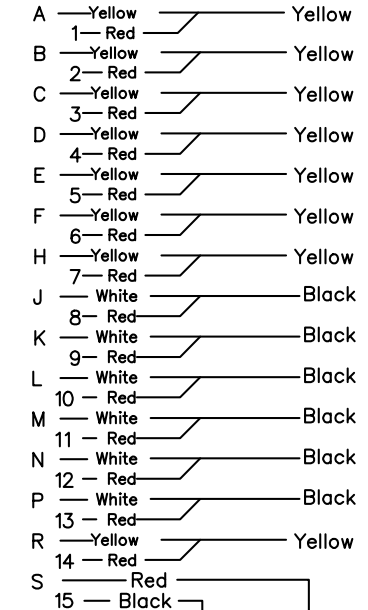


LEFT ENGINE (P3)
(G4 TWIN ONLY)



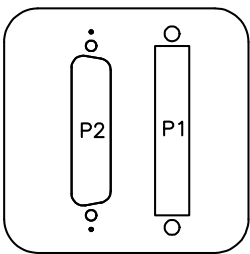
KEY

RIGHT ENGINE (P1)
(SINGLE ENGINE A/C)

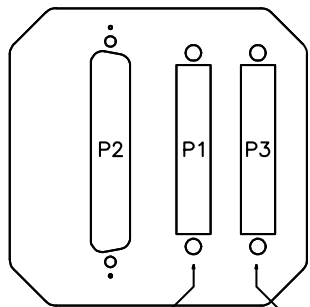


Sheath color

G1,G2,G3 REAR VIEW



G4 REAR VIEW



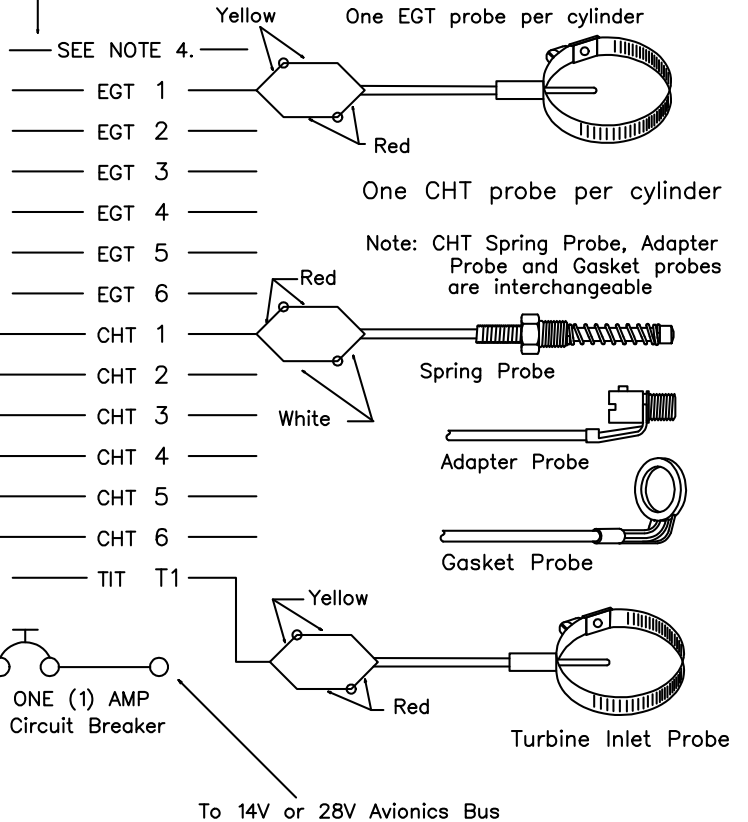
RIGHT ENGINE LEFT ENGINE

NOTES:

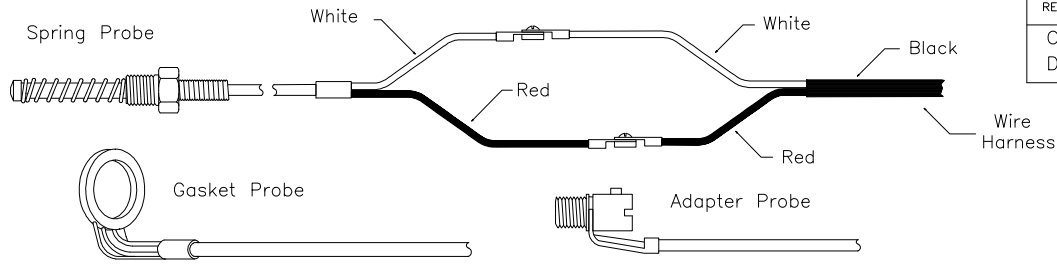
1. For 4 cylinder engines omit probe wires for cylinders 5 and 6.
2. For P2 connections see drawing no. 610C-315 or G4-315.
3. Power and ground wires 24 AWG.
4. TIT2 for G2, G3 and G4. Carburetor Temperature for G1.

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
B	ADDITION OF HARNESS DETAIL	930327	JBK
C	REVISED FOR CLARITY	960101	JWT
D	RE-ISSUED	080331	JY
E	ADD G1,G2,G3,G4 CONFIGURATIONS.	111101	JY

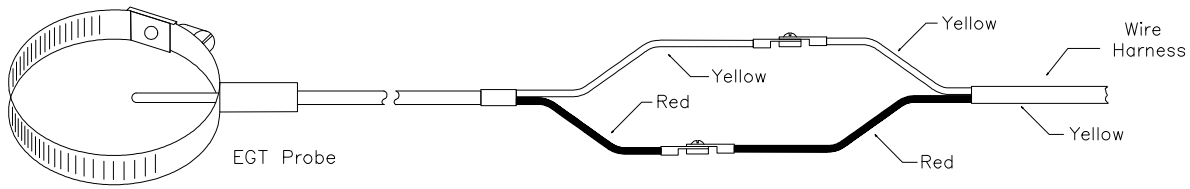
Harness wires are marked with cylinder number



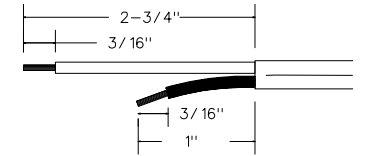
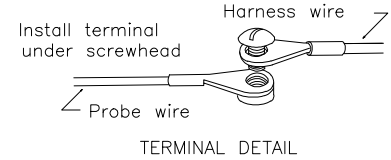
<h1>Insight</h1>	GEM WIRING DIAGRAM (G-Series EGT/CHT/TIT/CT WIRING)			
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2011	SIZE A	FILE NAME: G3WIREDE	DRAWING NO. 1200-018	DRN <i>JP</i>
SCALE NONE	DATE NOV 1/2011	SHEET 1 OF 1	CKD <i>JP</i>	REV E



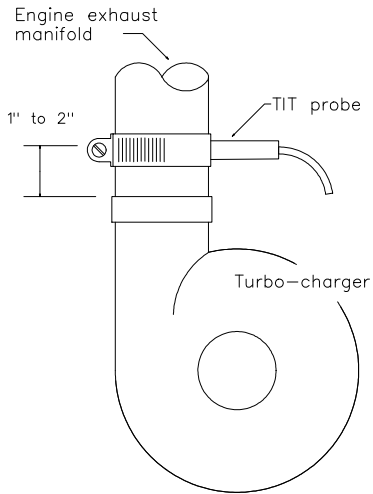
REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
C	RE-ISSUE	31 MAR 2008	JY
D	Changed EGT probe hole size.	1 JAN 2014	JY



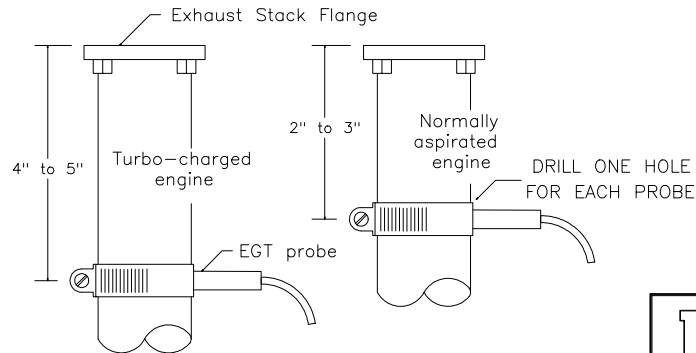
PROBE CONNECTION DIAGRAM



HARNESS WIRE STRIPPING DIMENSIONS



TURBO INLET PROBE LOCATION



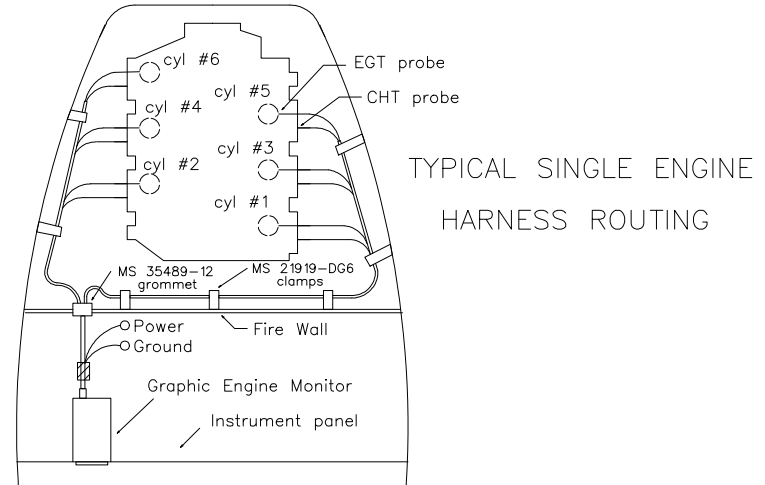
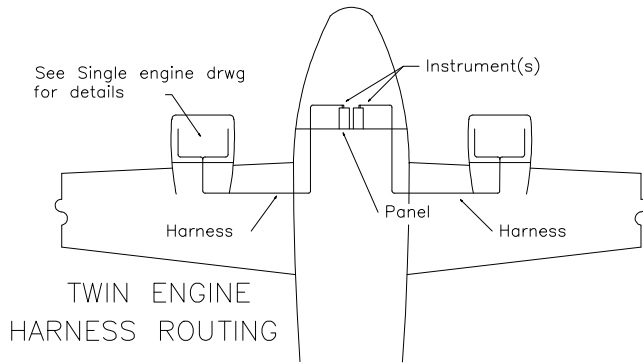
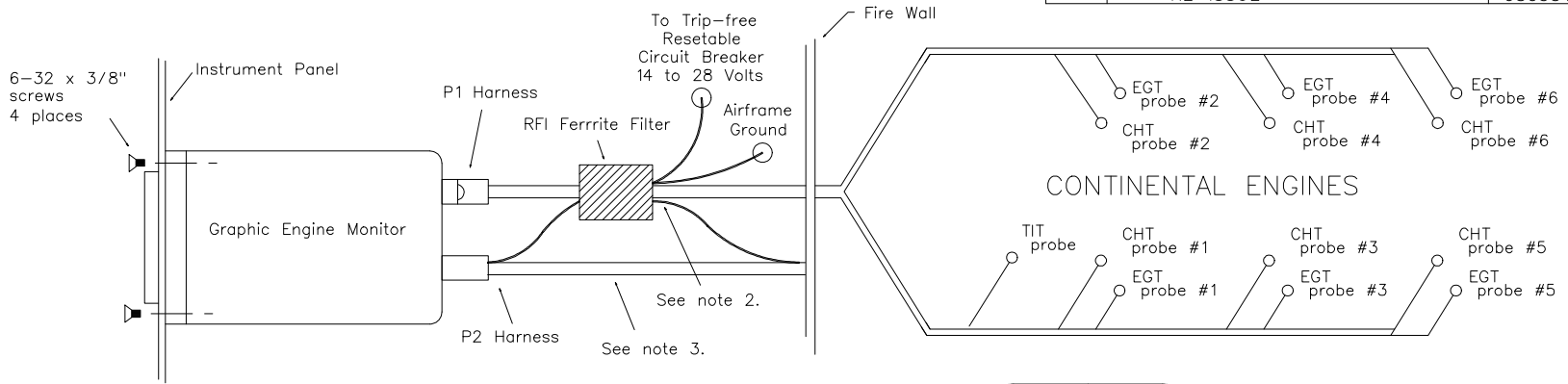
EGT PROBE LOCATIONS

PROBE HOLE SIZE (USE EITHER SIZE)	
DRILL BIT NUMBER	DIAMETER (INCHES)
30	0.1285"
29	0.136"

<h1>Insight</h1>	Graphic Engine Monitor PROBE CONNECTION						
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6						
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2008	SIZE A	FILE NAME: 8254D	DRAWING NO. 8254	DRN JY	CKD JY	QCD JY	REV D
SCALE None	DATE 1 JAN 2014	SHEET 1 OF 1					

PICTORIAL INSTRUMENT LAYOUT

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
E	RE-ISSUE	080331	JY



NOTES:

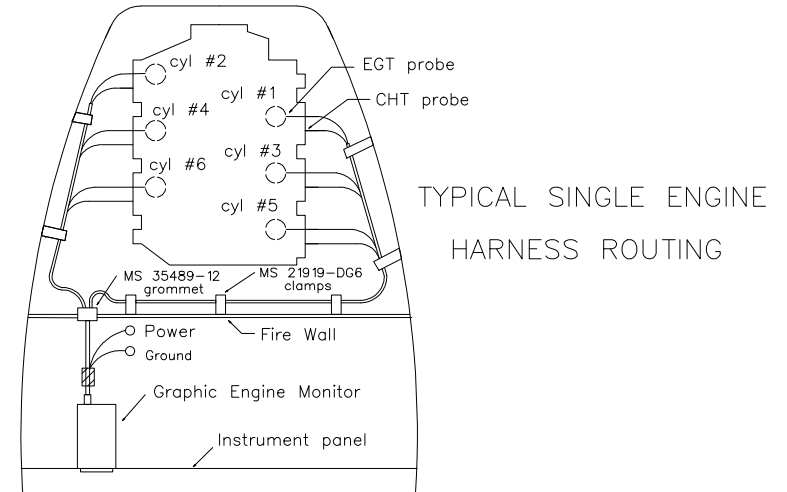
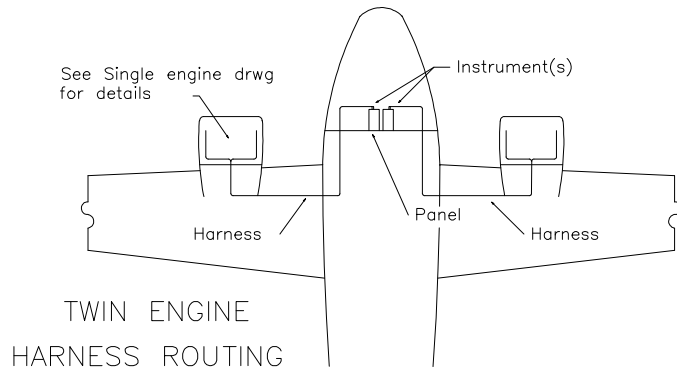
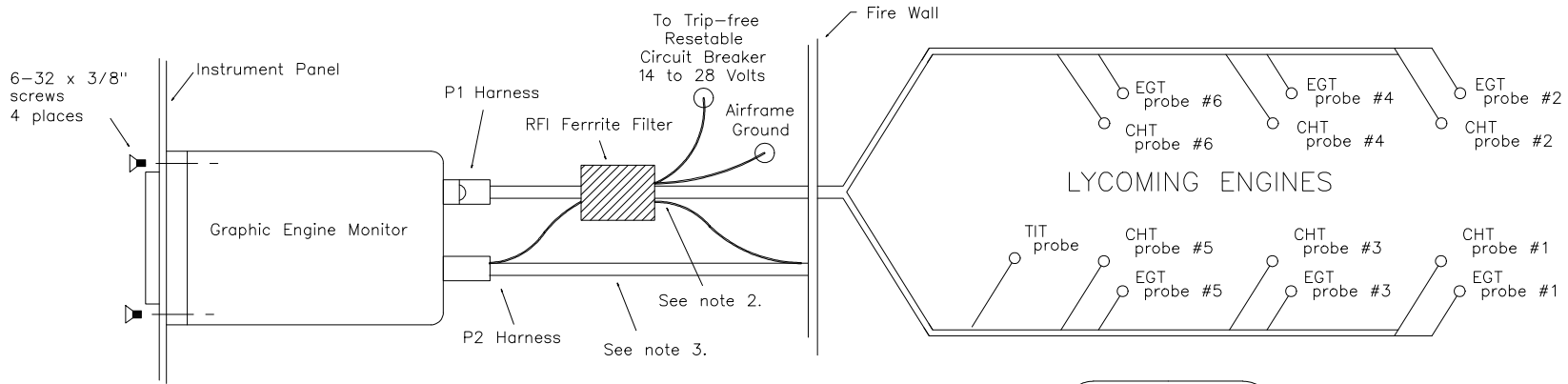
1. See drawing 8253 for Lycoming engines.
2. Route unshielded wires only through RFI ferrite filter.
3. Do not route shielded wires through RFI ferrite filter.
4. RFI ferrite filter may be solid bead or clamp-on type.

USE THIS DRAWING FOR CONTINENTAL ENGINES

Insight	Graphic Engine Monitor PROBE CONNECTION					
	Insight Avionic Inc. Box 122, Fort Erie, Ontario L2A 5M6					
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2008	SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD
	SCALE	8252E	8252	SP	SP	SP
	DATE		SHEET			
	080331		1 OF 1			

PICTORIAL INSTRUMENT LAYOUT

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
E	RE-ISSUE	080331	JY



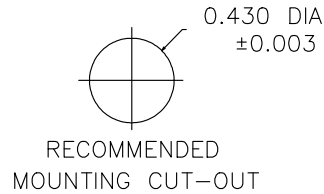
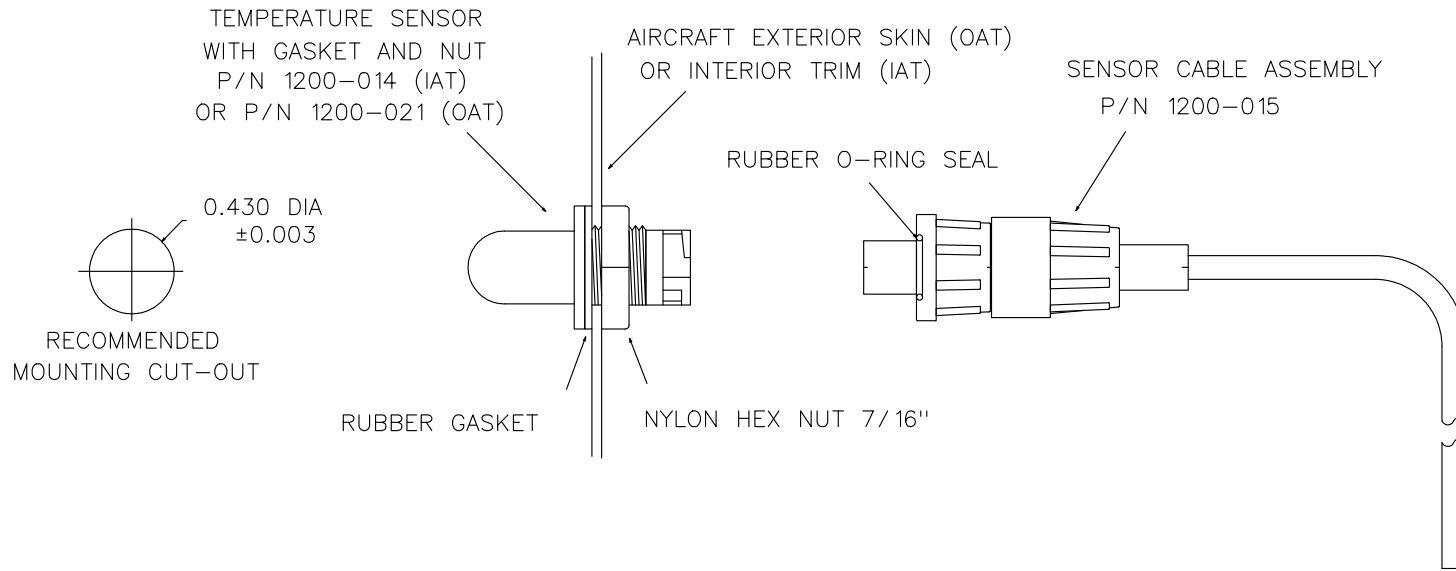
NOTES:

1. See drawing 8252 for Continental engines.
2. Route unshielded wires only through RFI ferrite filter.
3. Do not route shielded wires through RFI ferrite filter.
4. RFI ferrite filter may be solid bead or clamp-on type.

USE THIS DRAWING FOR LYCOMING ENGINES

Insight	Graphic Engine Monitor PROBE CONNECTION						
	Insight Avionic Inc. Box 122, Fort Erie, Ontario L2A 5M6						
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2008	SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
	A	8253E	8253	92	92	92	92
SCALE	DATE	SHEET					
none	080331	1 OF 1					

REVISIONS				
ZONE	REV	DESCRIPTION	DATE	APPROVED
	A	ADDED P/N 1200-021	930601	JWT
	B	ADDED NOTES	960101	JWT
	C	RE-ISSUE	080331	JY



NOTES:

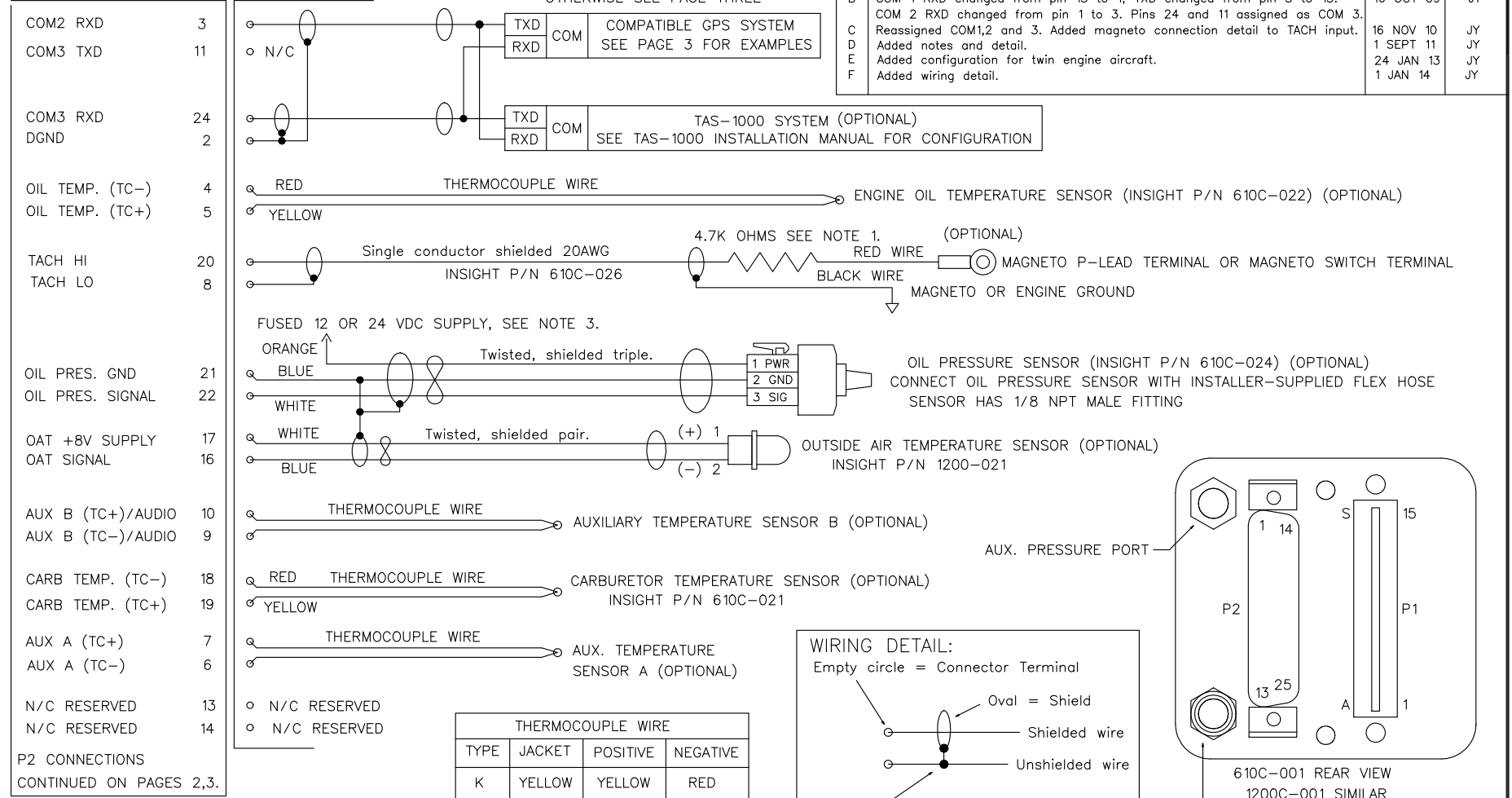
1. LOCATION OF OAT AND IAT PROBE IS CRITICAL TO CORRECT OPERATION REFER TO INSTRUCTIONS BEFORE LOCATING EITHER PROBE.
2. SENSOR CABLE ASSEMBLY IS PRE-TERMINATED BUT LENGTH MAY BE MODIFIED BY INSTALLER.
3. Torque nylon hex nut to 6 in.lb. MAX.
4. DO NOT apply any torque to bullet portion of sensor.

<h1>Insight</h1>	OUTSIDE AIR TEMPERATURE / INSIDE AIR TEMPERATURE PROBE MOUNTING DETAIL						
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6						
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2008	SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
		OATINSTC	1200-016	JY	JBK	JWT	C
SCALE	DATE	SHEET					
none	080331	1 OF 1					

610C-001 DISPLAY
OR 1200C-001 DISPLAY
P2 (D-SUB 25 RECEPTACLE) D-SUB 25 PLUG

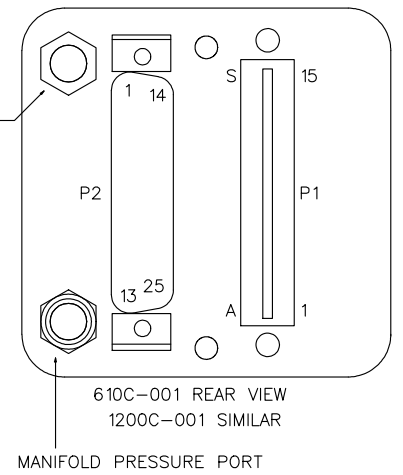
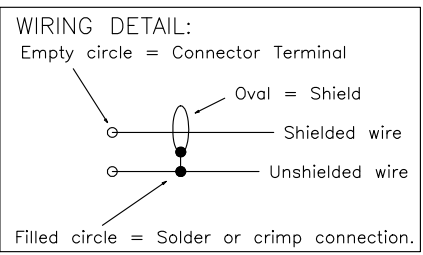
USE THIS CONFIGURATION TO CONNECT
A GPS AND TAS-1000 SYSTEM
OTHERWISE SEE PAGE THREE

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	See ECNs for changes.	31 MAR 08	JY
B	COM 1 RXD changed from pin 15 to 1, TXD changed from pin 3 to 15. COM 2 RXD changed from pin 1 to 3. Pins 24 and 11 assigned as COM 3.	16 OCT 09	JY
C	Reassigned COM1,2 and 3. Added magneto connection detail to TACH input.	16 NOV 10	JY
D	Added notes and detail.	1 SEPT 11	JY
E	Added configuration for twin engine aircraft.	24 JAN 13	JY
F	Added wiring detail.	1 JAN 14	JY



NOTES:

- IMPORTANT! USE INSIGHT CABLE P/N 610C-026 WITH 4,700 OHM POWER RESISTOR, ONE WATT (AS SUPPLIED).
- FOR FUEL FLOW AND VIBRATION SENSORS SEE PAGE TWO.
- Connect Oil Pressure Sensor Pin 1 to same +V supply as GEM P1-Harness, Pin S.
- Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
- Use pigtail connections where multiple wires connect to a single pin.



		DSUB 25-PIN INTERCONNECT DIAGRAM			
		Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2007	SIZE: G3in1F SCALE: NONE	FILE NAME: 610C-315 DATE: 1 JAN 2014	DRAWING NO.: 610C-315 DATE: 1 JAN 2014	DRN: [Signature] CKD: [Signature] QCD: [Signature] REV: F	SHEET: 1 OF 3

CONTINUED FROM PAGE 1

VIBRATION SENSOR, SLAVED GEM AND FUEL FLOW CONNECTIONS

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	See ECNs for changes.	31 MAR 08	JY
B	610C-001 pin 24 changed to COM 3 RVD, pin 11 changed to COM 3 TXD.	16 OCT 09	JY
C	1000-004A pin 2 changed to COM TXD, pin 7 changed to COM RVD. Reassigned COM 1 to FF/Vibration sensor.	16 NOV 10	JY
D	Added notes and detail.	1 SEPT 11	JY
E	Changed CPC from 7-pin to 6-pin.	24 JAN 13	JY
F	Revised notes for clarity.	1 JAN 14	JY

610C-001 DISPLAY
OR 1200C-001 DISPLAY
P2 (DSUB 25 SOCKET)

DSUB 25 PLUG

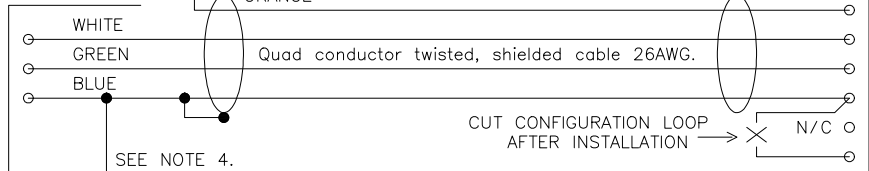
FUSED 12 OR 24 VDC SUPPLY, SEE NOTE 5.

RESISTOR ONLY FOR 24V AIRCRAFT
USE INSIGHT-SUPPLIED 750 OHM RESISTOR

6-PIN CPC
SOCKET

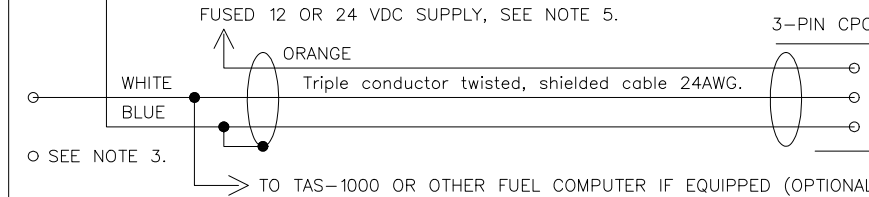
6-PIN CPC
PLUG

VIBRATION
COM 1 RVD 1
COM 1 TXD 15
DGND 25



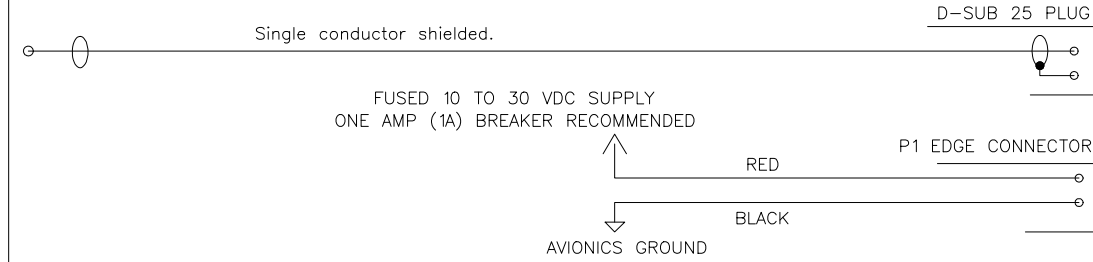
1	10 TO 30 VDC INPUT	VIBRATION SENSOR INSIGHT P/N 1000-004V OR INSIGHT P/N 1000-004A MOD. A
2	VIBE COM TXD	
6	VIBE COM RVD	
3	DGND	
5	NOT USED	
4	CONFIG LOOP	

FUEL FLOW
FF1 23
FF2 12



3	10 TO 30 VDC INPUT	FUEL FLOW SENSOR INSIGHT P/N 1000-004 FOR OTHERS SEE NOTE 7.
2	FF PULSE DATA OUT	
1	COMMON	

SLAVE
COM 2 TXD 14



SLAVE GEM DISPLAY (OPTIONAL)
MODEL G4-SINGLE OR G3 GEM

P/N 1200C-001 DISPLAY OR P/N 610C-001 DISPLAY	
P2 (D-SUB 25 RECEPTACLE)	
24	COM3 RXD
25	DGND
ALL OTHER PINS N/C	

P1 (30-PIN EDGE)	
S	10-30VDC INPUT
15	GROUND
ALL OTHER PINS N/C	

CONTINUED ON PAGE 3.

NOTES:

1. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
2. IF NO FF OR FF/V SENSORS ARE REQUIRED LEAVE PINS 1,12,15,23, AND 25 UNCONNECTED.
3. FF2 PIN 12 MAY CONNECT TO A RETURN FF SENSOR (1000-004). CONSULT INSIGHT.
4. Use pigtail connections where multiple wires connect to a single pin.
5. Connect FF and vibration sensor VDC inputs to same +V supply as GEM P1-harness, Pin S.
6. Use pigtail connection where multiple wires connect to a single pin.
7. Other fuel flow sensors are compatible, use adapter 610C-025, consult with Insight.

	DSUB 25-PIN INTERCONNECT DIAGRAM			
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2007	SIZE: G3in2F SCALE: NONE	FILE NAME: 610C-315 DATE: 1 JAN 14	DRAWING NO.: 610C-315 SHEET: 2 OF 3	DRN: [Signature] CKD: [Signature] QCD: [Signature] REV: F

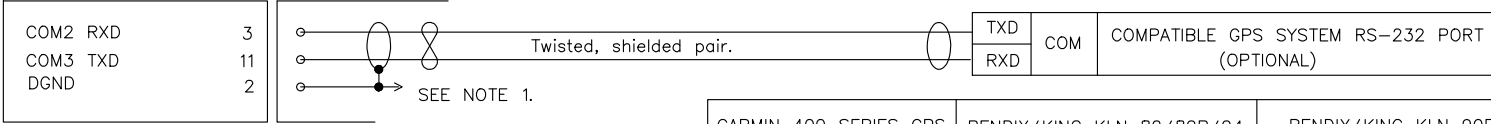
CONTINUED FROM PAGES 1,2.

CONNECTIONS TO GPS SYSTEM IN SINGLE AND TWIN ENGINE AIRCRAFT WITHOUT A TAS-1000 SYSTEM

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
E	See ECNs for changes.	24 JAN 13	JY
F	Revised notes for clarity.	1 JAN 14	JY

MODEL G3 (610C-001) OR G4-Single (1200C-001) IN A SINGLE ENGINE AIRCRAFT

P2 (D-SUB 25 RECEPTACLE) D-SUB 25 PLUG

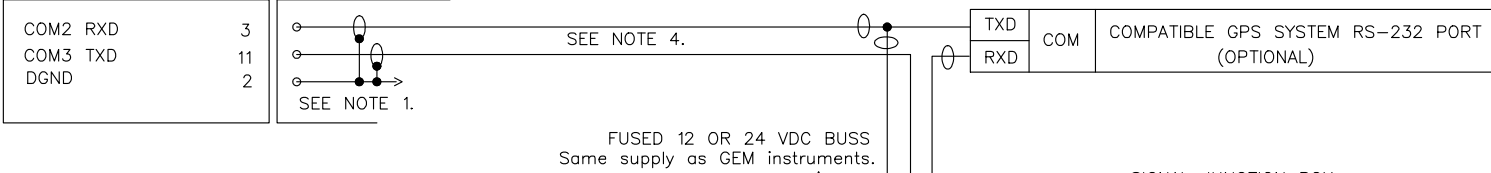


GARMIN 400 SERIES GPS (P4001) 56 RS232 OUT (P4001) 57 RS232 IN	BENDIX/KING KLN 89/89B/94 (P891) 2 RS232 OUT (P891) 1 RS232 IN	BENDIX/KING KLN 90B (J901) 13 RS232 OUT (J901) 36 RS232 IN	UPS APOLLO GX50/55/60/65 37 PIN CONNECTOR PIN 5 RS232 OUT PIN 4 RS232 IN
--	--	--	---

LEFT ENGINE GEM INSTRUMENT IN A TWIN ENGINE AIRCRAFT

MODEL G3 (610C-001) OR G4-Single (1200C-001)

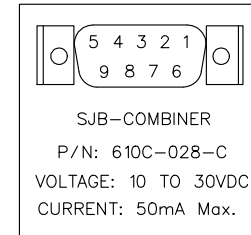
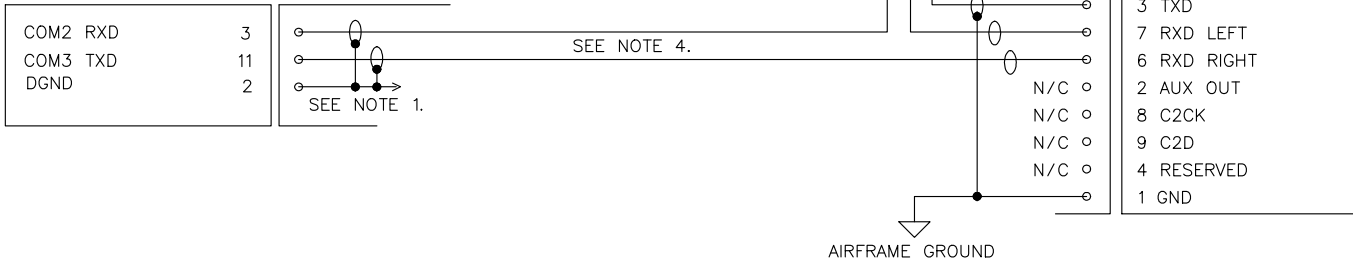
P2 (D-SUB 25 RECEPTACLE) D-SUB 25 PLUG



RIGHT ENGINE GEM INSTRUMENT IN A TWIN ENGINE AIRCRAFT

MODEL G3 (610C-001) OR G4-Single (1200C-001)

P2 (D-SUB 25 RECEPTACLE) D-SUB 25 PLUG



NOTES:

1. GROUND PIN SHARED WITH OTHER RS232 DEVICES, SEE PAGE ONE.
2. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
3. Use pigtail connections where multiple wires connect to a single pin.
4. Single conductor shielded cable recommended for long runs.

<h1>Insight</h1>	DSUB 25-PIN INTERCONNECT DIAGRAM			
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2013	SIZE: G3in3F	FILE NAME: 610C-315	DRAWING NO. 610C-315	DRN: JY CKD: JY QCD: JY REV: F
SCALE: NONE	DATE: 1 JAN 2014	SHEET: 3 OF 3		

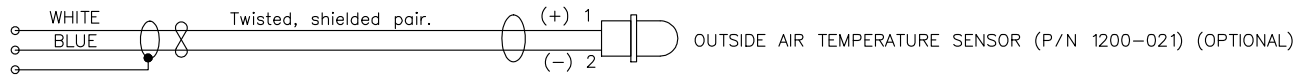
MODEL G4-TWIN OR G9 GEM
 P/N 1200C-001 DISPLAY
 P2 (D-SUB 37 RECEPTACLE) D-SUB 37 PLUG

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	First issue.	10 DEC 10	JY
B	Added Page Three for GPS and TAS-1000.	24 JAN 13	JY
C	Added notes for G9 single-engine installations.	1 OCT 13	JY

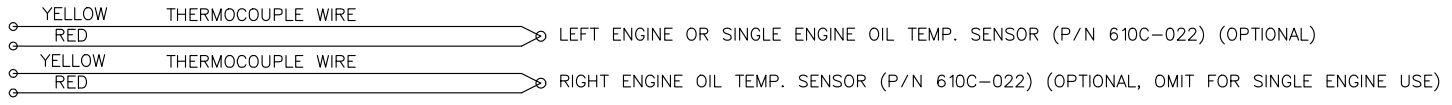
N/C RESERVED 25
 COM3 RXD 14
 DGND 33
 COM2 RXD 24
 COM3 TXD 32



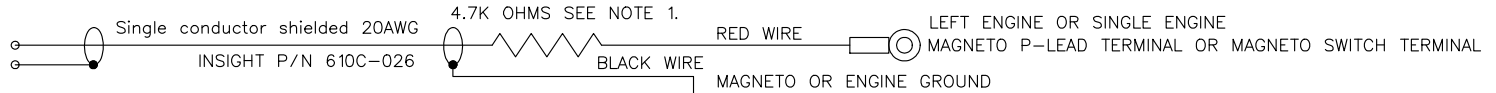
OAT +8V SUPPLY 28
 OAT SIGNAL 9
 OAT SHIELD 27



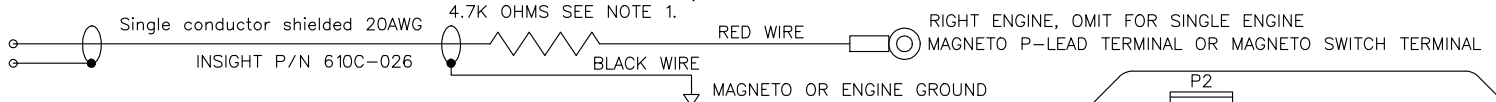
OIL TEMP. 1 (TC+) 26
 OIL TEMP. 1 (TC-) 8
 OIL TEMP. 2 (TC+) 2
 OIL TEMP. 2 (TC-) 21



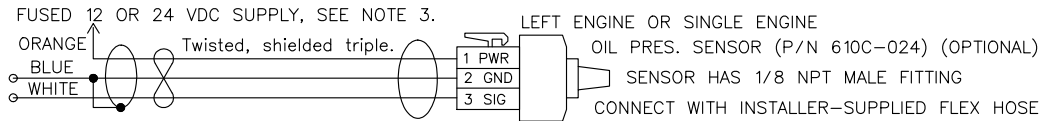
TACH 1 HI 36
 TACH 1 LO 18



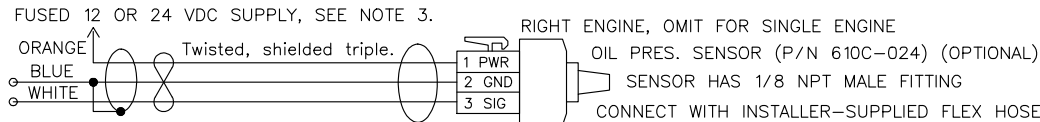
TACH 2 HI 19
 TACH 2 LO 37



OIL PRES. 1 GND 34
 OIL PRES. 1 SIGNAL 15



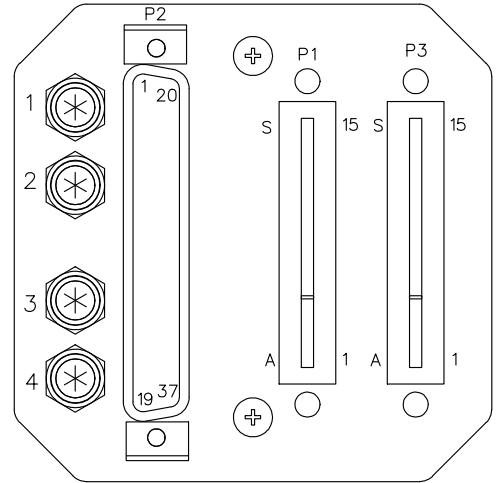
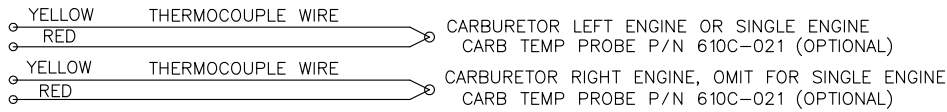
OIL PRES. 2 GND 20
 OIL PRES. 2 SIGNAL 1



AUX 1 (TC+) 29
 AUX 1 (TC-) 10

o N/C
 o N/C

CARB 1 (TC+) 7
 CARB 1 (TC-) 6
 CARB 2 (TC+) 3
 CARB 2 (TC-) 22



CONTINUED ON NEXT PAGE

NOTES:

- IMPORTANT! USE INSIGHT CABLE P/N 610C-026 WITH 4,700 OHM POWER RESISTOR, ONE WATT (AS SUPPLIED).
- FOR FUEL FLOW AND VIBRATION SENSORS SEE NEXT PAGE.
- Connect Oil Pressure Sensor Pin 1 to same +V supply as GEM P1-Harness, Pin S.
- Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.

THERMOCOUPLE WIRE			
TYPE	JACKET	POSITIVE	NEGATIVE
K	YELLOW	YELLOW	RED
J	BLACK	WHITE	RED



G4/G9 DSUB 37-PIN INTERCONNECT DIAGRAM

Insight Avionics Inc.
 Box 122, Fort Erie, Ontario L2A 5M6

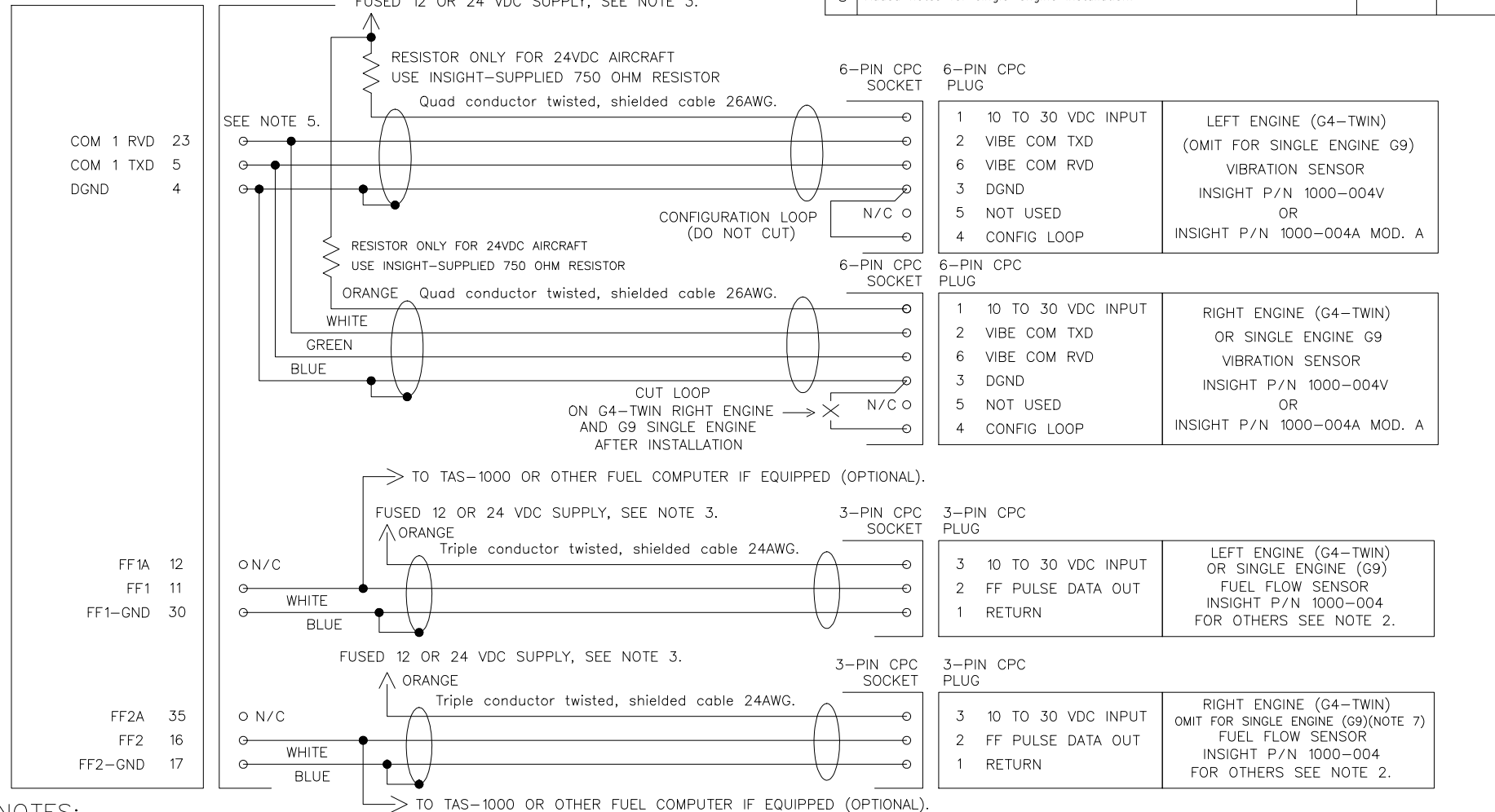
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SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
	G4in1C	G4-315	JWT	98	98	C
SCALE	DATE	SHEET				
NONE	1 OCT 13	1 OF 3				

CONTINUED FROM PAGE 1

MODEL G4-TWIN OR G9 GEM
P/N 1200C-001 DISPLAY
P2 (DSUB 37 SOCKET) DSUB 37 PLUG

FUEL FLOW AND VIBRATION SENSOR WIRING FOR G4 TWIN AND G9 SINGLE



COM 1 RVD 23
COM 1 TXD 5
DGND 4

FF1A 12
FF1 11
FF1-GND 30

FF2A 35
FF2 16
FF2-GND 17

NOTES:

1. Use 24 AWG copper wire with TEFLON or equivalent insulation unless otherwise specified.
2. For connection to other fuel flow sensors use P/N 610C-025, see Drawing no. 101116A.
3. Connect FF and vibration sensor VDC inputs to same +V supply as GEM P1-harness, Pin S.
4. Note that a 750 Ohm resistor is required for each vibration sensor in 24 Volt aircraft.
5. Use pigtail connections where multiple wires connect to a single pin.
6. Refer to TAS-1000 Installation Manual for pin-numbers.
7. May be configured for single engine return fuel line, contact Insight.

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	FIRST ISSUE	10 DEC 10	JY
B	Changed vibration sensor connector from 7 to 6 pins.	24 JAN 13	JY
C	Added notes for single-engine installation.	1 OCT 13	JY

1	10 TO 30 VDC INPUT	LEFT ENGINE (G4-TWIN)
2	VIBE COM TXD	(OMIT FOR SINGLE ENGINE G9)
3	VIBE COM RVD	VIBRATION SENSOR
4	DGND	INSIGHT P/N 1000-004V
5	NOT USED	OR
6	CONFIG LOOP	INSIGHT P/N 1000-004A MOD. A

1	10 TO 30 VDC INPUT	RIGHT ENGINE (G4-TWIN)
2	VIBE COM TXD	OR SINGLE ENGINE G9
3	DGND	VIBRATION SENSOR
4	CONFIG LOOP	INSIGHT P/N 1000-004V
5	NOT USED	OR
6	VIBE COM RVD	INSIGHT P/N 1000-004A MOD. A

3	10 TO 30 VDC INPUT	LEFT ENGINE (G4-TWIN)
2	FF PULSE DATA OUT	OR SINGLE ENGINE (G9)
1	RETURN	FUEL FLOW SENSOR
		INSIGHT P/N 1000-004
		FOR OTHERS SEE NOTE 2.

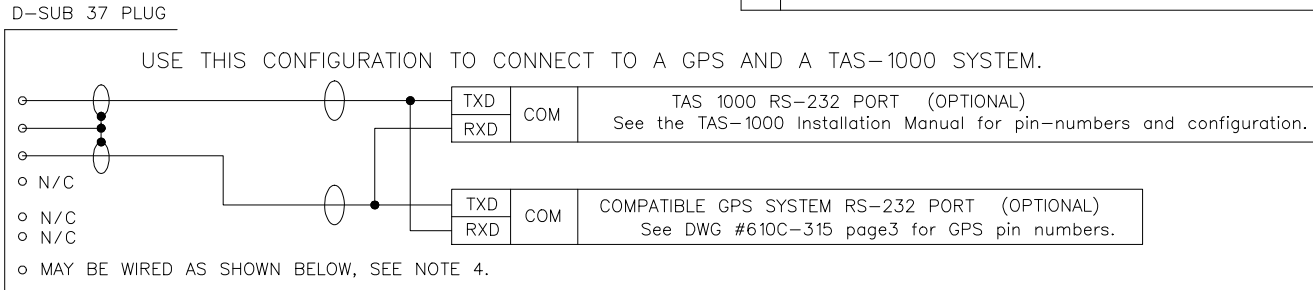
3	10 TO 30 VDC INPUT	RIGHT ENGINE (G4-TWIN)
2	FF PULSE DATA OUT	OMIT FOR SINGLE ENGINE (G9)(NOTE 7)
1	RETURN	FUEL FLOW SENSOR
		INSIGHT P/N 1000-004
		FOR OTHERS SEE NOTE 2.

	G4/G9 DSUB 37-PIN INTERCONNECT DIAGRAM						
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6						
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2010	SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
	SCALE	G4in2C	G4-315				C
	DATE	1 OCT 13	SHEET	2 OF 3			

GPS AND AIR-DATA COMPUTER WIRING
AND OPTIONAL MASTER-SLAVE GEM WIRING
FOR G4-TWIN ENGINE AND G9 SINGLE ENGINE

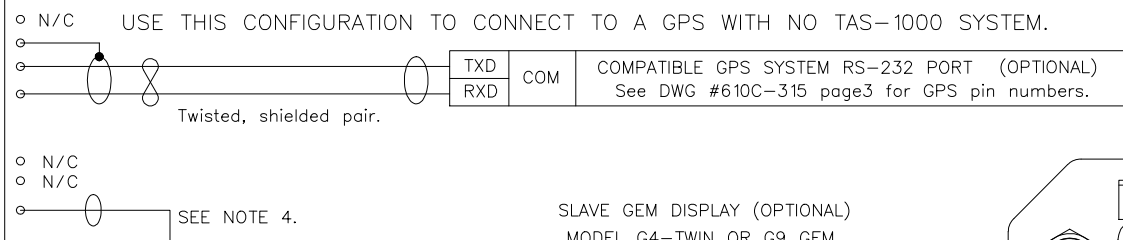
MODEL G4-TWIN OR G9 GEM
P/N 1200C-001 DISPLAY
P2 (D-SUB 37 RECEPTACLE) D-SUB 37 PLUG

COM3 RXD	14
DGND	33
COM2 RXD	24
COM3 TXD	32
RESERVED A	31
RESERVED B	13
COM2 TXD	25



P2 (D-SUB 37 RECEPTACLE) D-SUB 37 PLUG

COM3 RXD	14
DGND	33
COM2 RXD	24
COM3 TXD	32
RESERVED A	31
RESERVED B	13
COM2 TXD	25



SLAVE GEM DISPLAY (OPTIONAL)
MODEL G4-TWIN OR G9 GEM

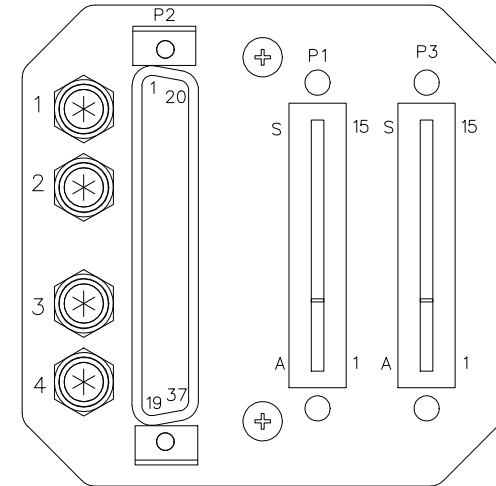
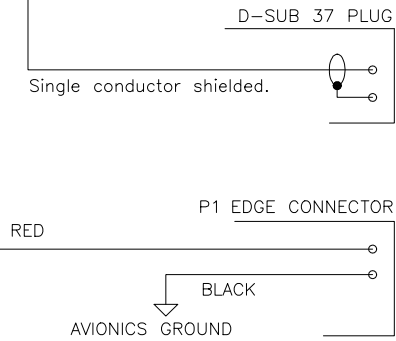
P/N 1200C-001 DISPLAY
P2 (D-SUB 37 RECEPTACLE)

14	COM3 RXD
33	DGND
ALL OTHER PINS N/C	

P1 (30-PIN EDGE)

S	10-30VDC INPUT
15	GROUND
ALL OTHER PINS N/C	

FUSED 10 TO 30 VDC SUPPLY
ONE AMP (1A) BREAKER RECOMMENDED



NOTES:

1. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
2. Refer to the TAS-1000 Installation Manual for pin numbers and configuration.
3. Refer to the GPS manufacturers data for pin numbers and configuration.
4. Wire as shown for a slaved GEM installation. Otherwise leave pin 25 N/C.

Insight

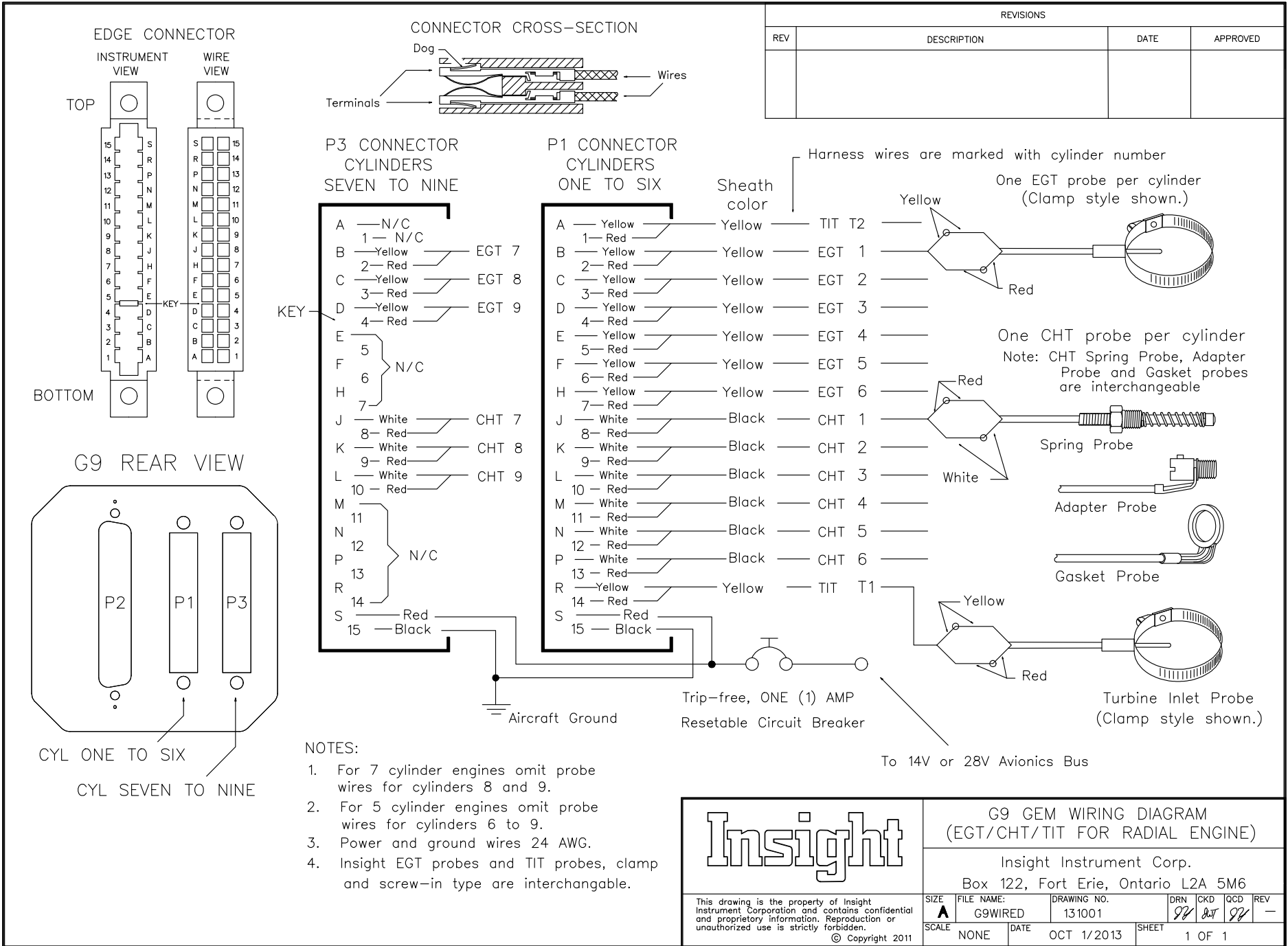
G4/G9 DSUB 37-PIN INTERCONNECT DIAGRAM

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SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
	G4in3C	G4-315	JWT	92	92	C
SCALE	DATE	SHEET				
NONE	1 OCT 13	3 OF 3				

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	First issue.	10 DEC 10	JY
B	Added notes for GPS.	24 JAN 13	JY
C	Added notes for G9 single engine and Master-Slave GEM.	1 OCT 13	JY



REVISIONS			
REV	DESCRIPTION	DATE	APPROVED

- NOTES:
- For 7 cylinder engines omit probe wires for cylinders 8 and 9.
 - For 5 cylinder engines omit probe wires for cylinders 6 to 9.
 - Power and ground wires 24 AWG.
 - Insight EGT probes and TIT probes, clamp and screw-in type are interchangeable.

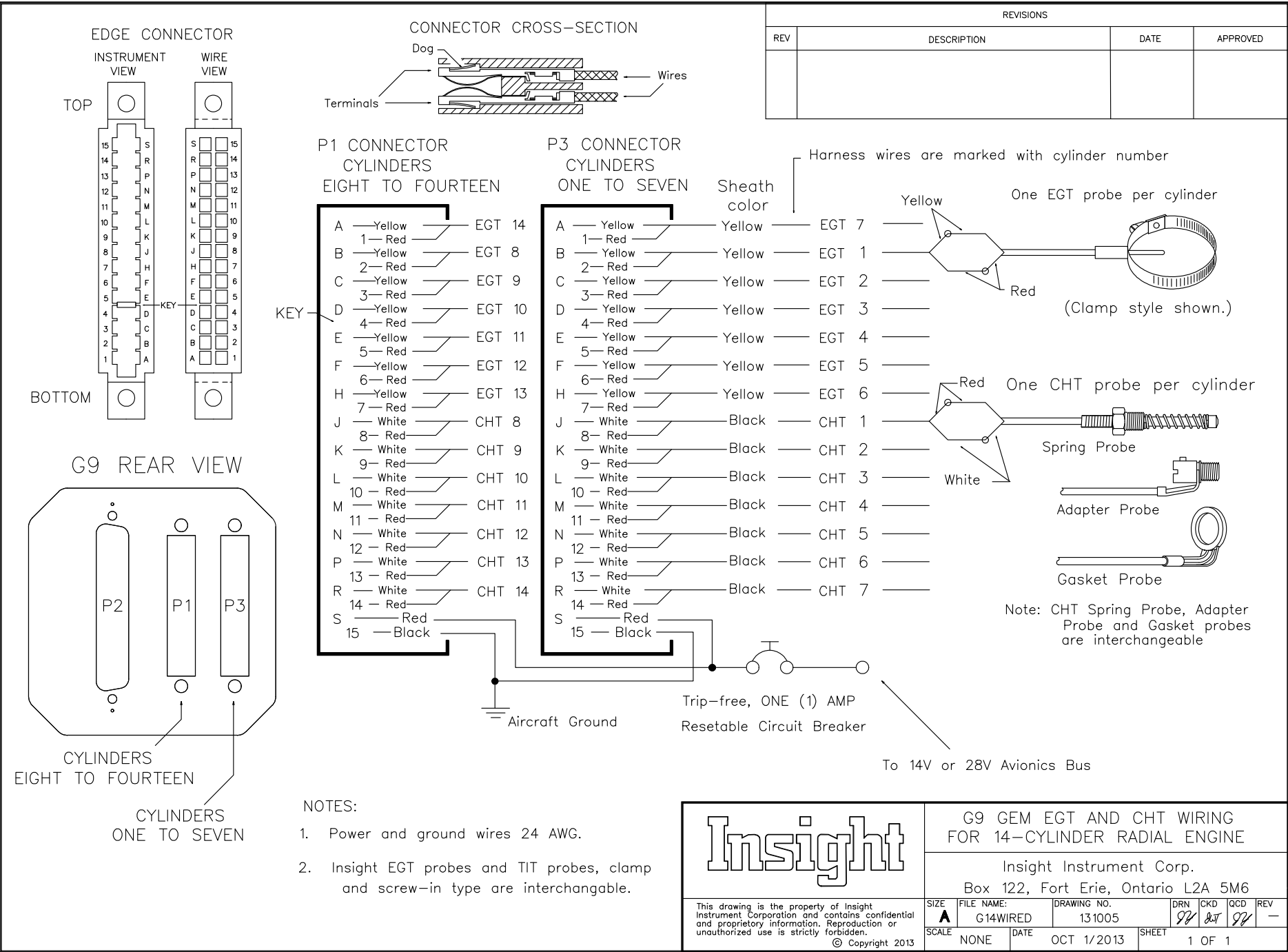
Insight

G9 GEM WIRING DIAGRAM
(EGT/CHT/TIT FOR RADIAL ENGINE)

Insight Instrument Corp.
Box 122, Fort Erie, Ontario L2A 5M6

SIZE A	FILE NAME: G9WIRED	DRAWING NO. 131001	DRN <i>JK</i>	CKD <i>JK</i>	QC'D <i>JK</i>	REV —
SCALE NONE	DATE OCT 1/2013	SHEET 1 OF 1				

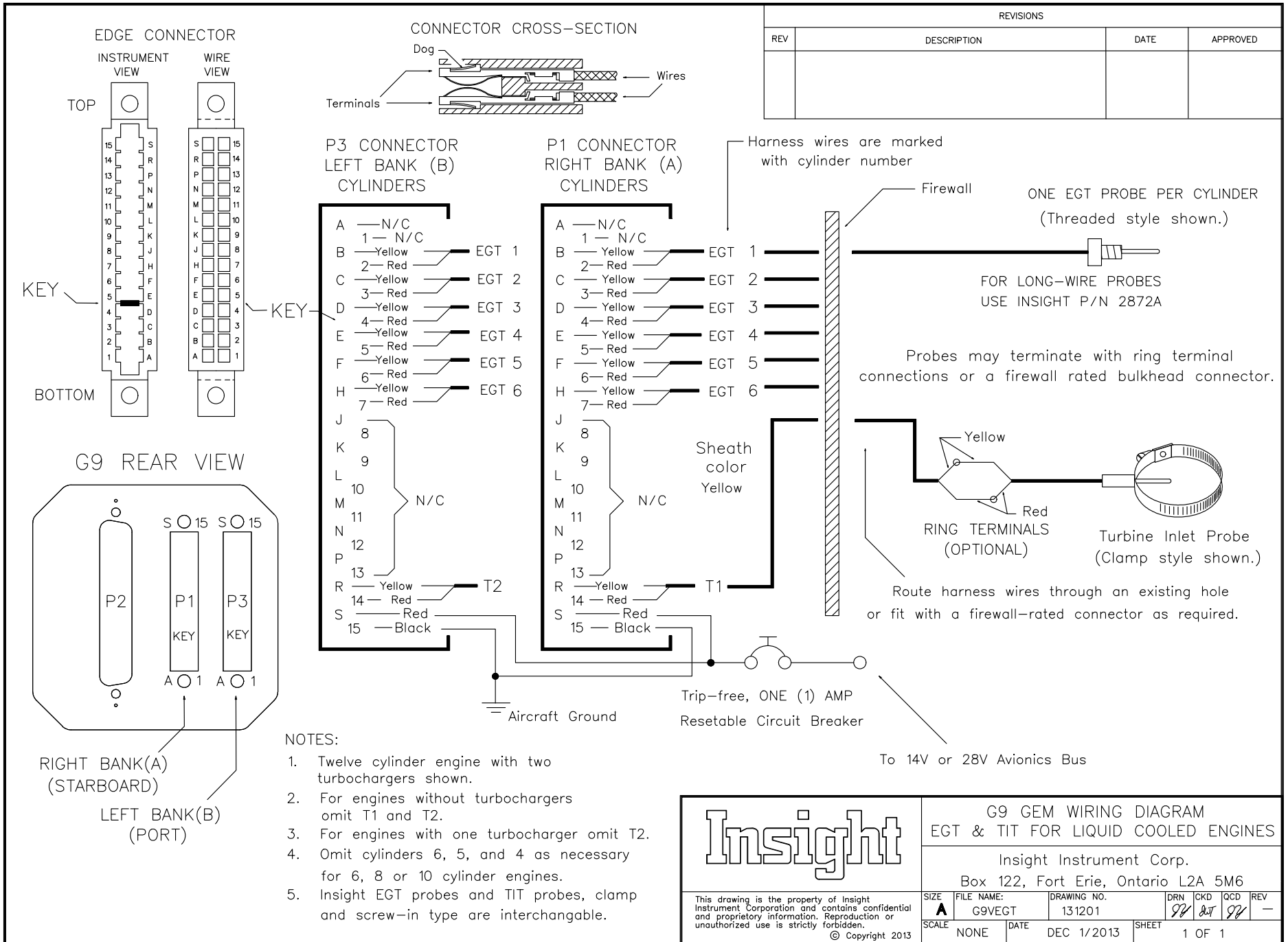
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REVISIONS			
REV	DESCRIPTION	DATE	APPROVED

- NOTES:
1. Power and ground wires 24 AWG.
 2. Insight EGT probes and TIT probes, clamp and screw-in type are interchangeable.

		G9 GEM EGT AND CHT WIRING FOR 14-CYLINDER RADIAL ENGINE			
		Insight Instrument Corp. Box 122, Fort Erie, Ontario L2A 5M6			
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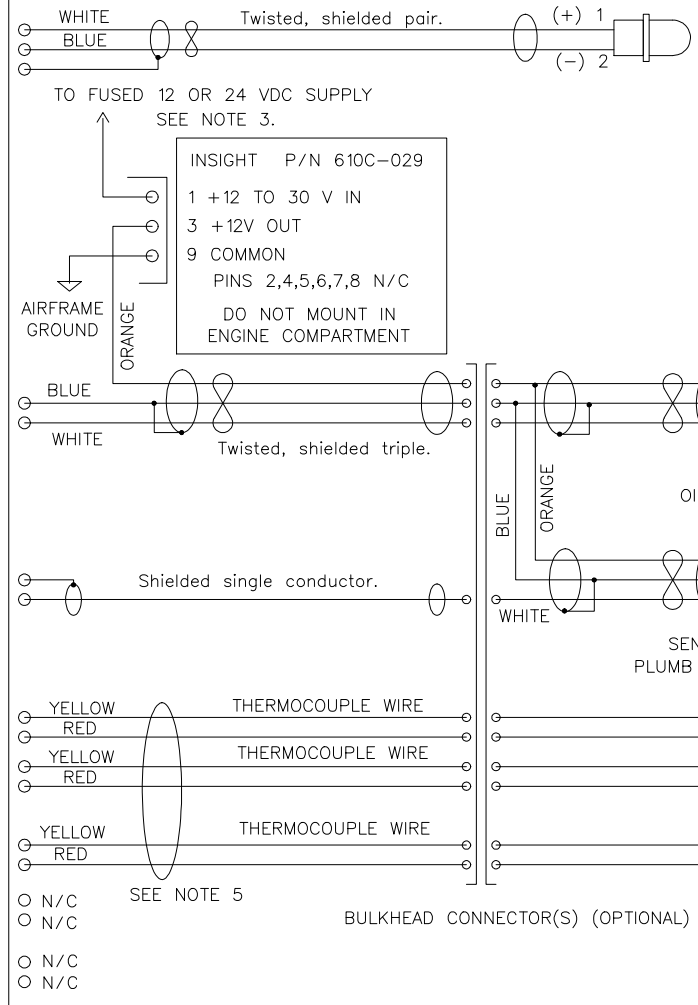
REVISIONS			
REV	DESCRIPTION	DATE	APPROVED

<h1>Insight</h1>	G9 GEM WIRING DIAGRAM EGT & TIT FOR LIQUID COOLED ENGINES			
	Insight Instrument Corp. Box 122, Fort Erie, Ontario L2A 5M6			
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SCALE NONE	DATE DEC 1/2013	SHEET 1 OF 1	QCD JK	REV —

INSIGHT G9 GEM DISPLAY
P2 (D-SUB 37 RECEPTACLE) D-SUB 37 PLUG

OUTSIDE AIR TEMPERATURE SENSOR
(P/N 1200-021) (OPTIONAL)

OAT +8V SUPPLY	28
OAT SIGNAL	9
OAT SHIELD	27
FUEL PRES. GND	20
FUEL PRES. SIGNAL	1
OIL PRES. GND	34
OIL PRES. SIGNAL	15
OIL TEMP. (TC+)	26
OIL TEMP. (TC-)	8
COOLANT TEMP. (TC+)	2
COOLANT TEMP. (TC-)	21
CARB 1 (TC+)	7
CARB 1 (TC-)	6
CARB 2 (TC+)	3
CARB 2 (TC-)	22
AUX 1 (TC+)	29
AUX 1 (TC-)	10



TO FUSED 12 OR 24 VDC SUPPLY
SEE NOTE 3.

INSIGHT P/N 610C-029

1 +12 TO 30 V IN

3 +12V OUT

9 COMMON

PINS 2,4,5,6,7,8 N/C

DO NOT MOUNT IN ENGINE COMPARTMENT

AIRFRAME GROUND

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED

FUEL PRES. SENSOR (OPTIONAL)
5/8-18 MOUNTING THREADS
SENSOR HAS 1/8 NPT FEMALE FITTING
PLUMB WITH INSTALLER-SUPPLIED FLEX HOSE

SHIELD INTERNALLY CONNECTED TO COMMON

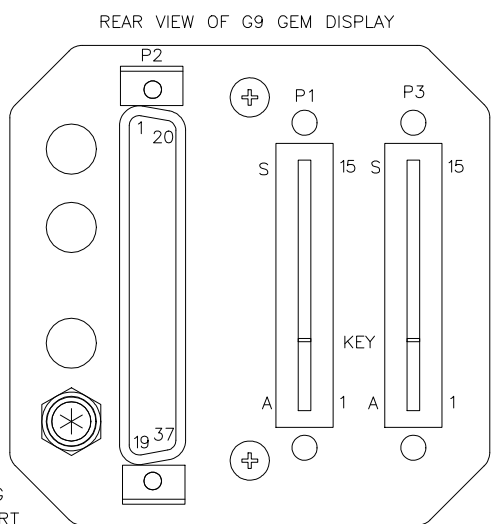
OIL PRES. SENSOR (OPTIONAL)
(INSIGHT P/N 610C-024)

SENSOR HAS 1/8 NPT MALE FITTING
PLUMB WITH INSTALLER-SUPPLIED FLEX HOSE

OIL TEMP. SENSOR
(P/N 610C-022) (OPTIONAL)

COOLANT TEMP. SENSOR
(P/N 610C-022) (OPTIONAL)

CARB TEMP PROBE (OPTIONAL)
INSIGHT P/N 610C-021



INTERNAL SENSOR
#2 FLARE MALE FITTING
MANIFOLD PRESSURE PORT

NOTES:

- ROUTE WIRING THROUGH EXISTING HOLE(S), OR INTERCONNECT THROUGH BULKHEAD CONNECTOR(S) AS REQUIRED.
- FOR FUEL FLOW AND ADDITIONAL CONNECTIONS CONTACT INSIGHT.
- Connect Pressure Sensor Supplies to same +V supply as GEM P1-Harness, Pin S.
- Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
- Route thermocouple wires through ferrite RF filter if required.

Insight

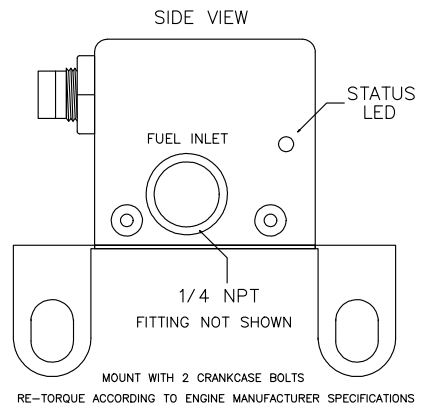
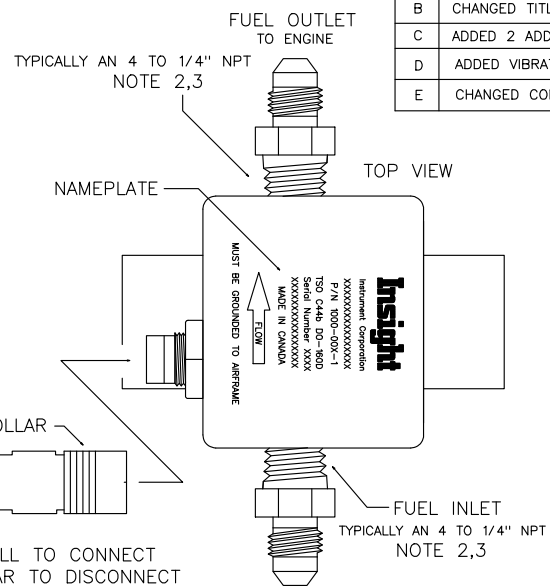
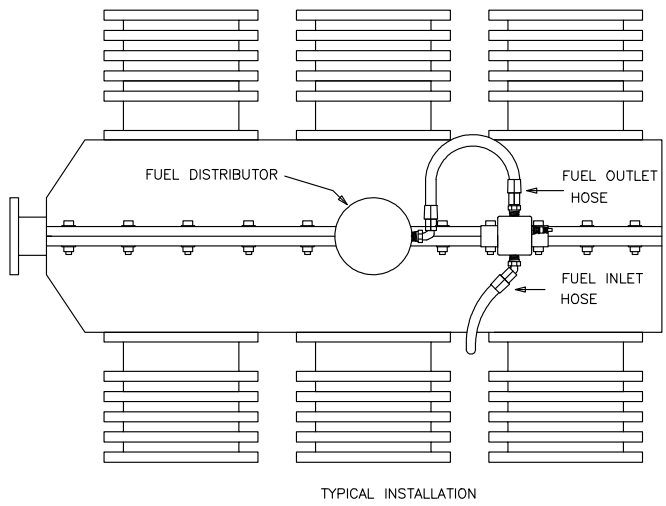
G9 DSUB 37-PIN INTERCONNECT DIAGRAM
FOR LIQUID COOLED ENGINES

Insight Instrument Corp.
Box 122, Fort Erie, Ontario L2A 5M6

SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
SCALE	G9VIN1	131202	<i>[Signature]</i>	<i>[Signature]</i>	<i>[Signature]</i>	-
NONE	DATE	1 OCT 13	SHEET	1 OF 1		

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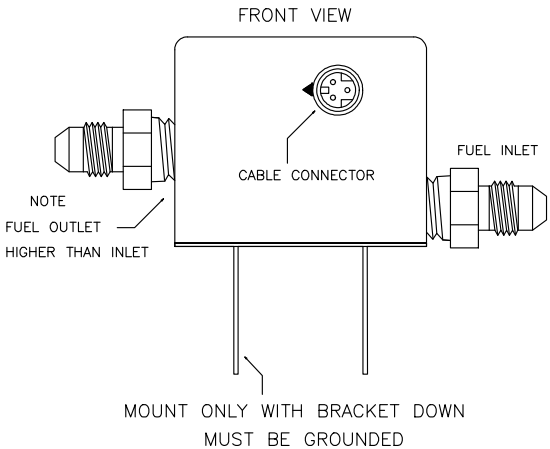
REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	Added housing and bracket.	June 22/00	JWT
B	CHANGED TITLE	June 15/01	R.R.
C	ADDED 2 ADDITIONAL MOUNTING OPTIONS	MAY 26/03	R.R.
D	ADDED VIBRATION AND K-FACTOR OPTIONS	DEC 10/10	JY
E	CHANGED CONNECTOR.	NOV 30/12	JY



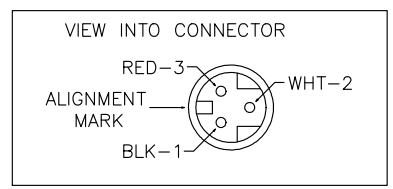
PART NUMBERS			
	INJECTOR	CARBURETOR	
FLOW SENSOR	1000-004-1(-Knnnn)(-EPR)	1000-004-4(-Knnnn)(-EPR)	
TRUE FLOW SENSOR	1000-006-1X SEE NOTE 12.	1000-006-4X	SEE NOTE 12.

NOTES:

1. Install INSIGHT FLOW SENSOR, one per engine, TWO per engine for pressure carburetor.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use ONLY approved fuel line assemblies.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied electrical harness with 3-pin connector, P/N 1000-809.
6. A straight length of fuel line upstream from the sensor inlet will improve accuracy by reducing turbulence in the rotor.
7. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP.
8. Mount transducer with NAMEPLATE facing up.
9. Confirm all wiring PRIOR to powering up.
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MANUFACTURE AND MODEL WHEN ORDERING TRUE FLOW SENSOR, BENDIX KING B-FORMAT, APOLLO AND GARMIN Z-FORMAT.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Knnnn.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED, SEE WIRING DIAGRAM.

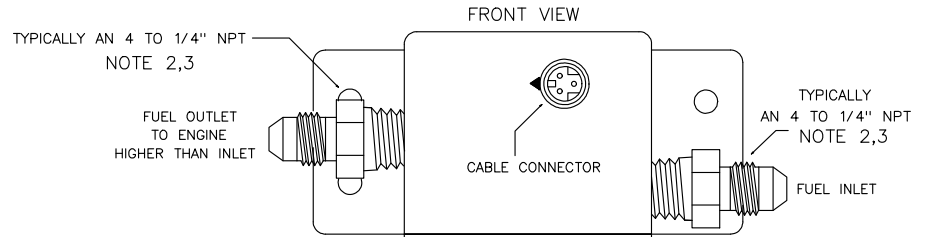
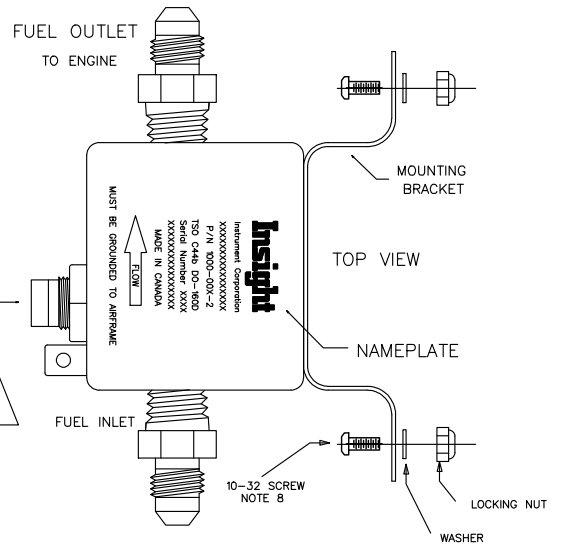
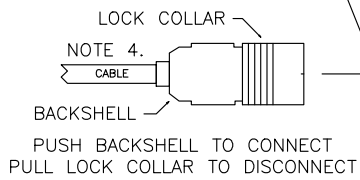
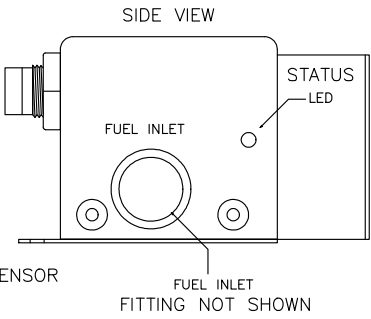
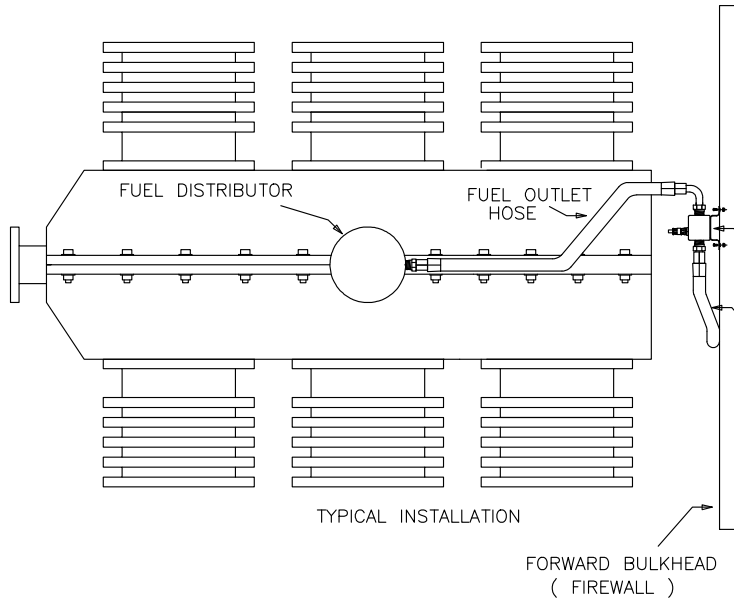


PIN	FUNCTION	WIRE COLOUR
1	COMMON (DO NOT GROUND)	BLACK OR WHITE/BLUE
2	SIGNAL OUT	WHITE
3	10 TO 30 VDC INPUT	RED OR WHITE/ORANGE



		FLOW SENSOR AND TRUE FLOW SENSOR MOUNTING DIAGRAM (BRACKET TYPE 1)				
		Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6				
SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
	fmount1E	1000-303	JWT	92	92	E
SCALE	DATE	SHEET	REV			
none	June 1/99	1 OF 3				

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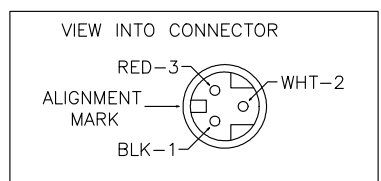


PART NUMBERS			
	INJECTOR	CARBURETOR	
FLOW SENSOR	1000-004-2(-Knnnnn)(-EPR)	1000-004-5(-Knnnnn)(-EPR)	
TRUE FLOW SENSOR	1000-006-2X SEE NOTE 12.	1000-006-5X	SEE NOTE 12.

NOTES:

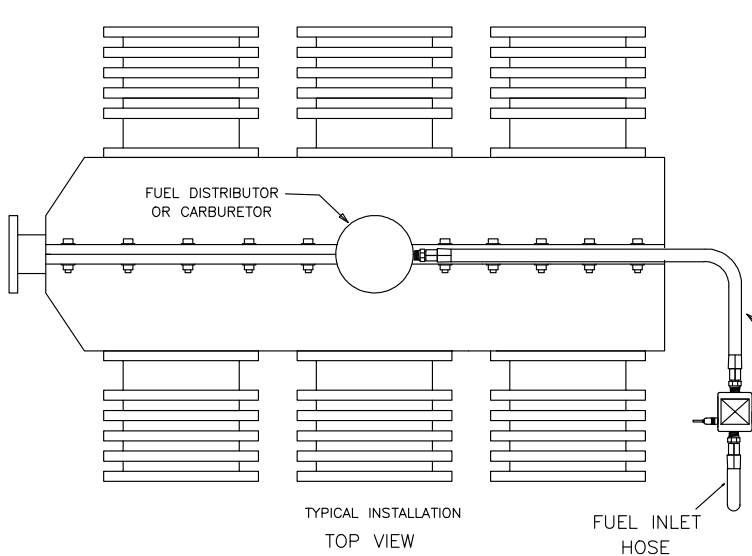
1. Install INSIGHT FLOW SENSOR, one per engine, TWO per engine for pressure carburetor.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied. Use appropriate fittings to match airplane fuel lines.
3. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
4. INSIGHT supplied electrical harness with 3-pin connector, P/N 1000-809.
5. A straight length of fuel line upstream from the sensor inlet will improve accuracy by reducing turbulence in the rotor.
6. Mount on firewall or airframe with NAMEPLATE facing up.
7. Installer may drill additional holes in bracket if necessary.
8. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP.
9. Confirm all wiring PRIOR to powering up.
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MODEL WHEN ORDERING TRUE FLOW SENSOR, B = BENDIX KING, Z = APPOLLO AND GARMIN.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Knnnnn.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED, SEE WIRING DIAGRAM.

PIN	FUNCTION	WIRE COLOUR
1	COMMON (DO NOT GROUND)	BLACK OR WHITE/BLUE
2	SIGNAL OUT	WHITE
3	10 TO 30 VDC INPUT	RED OR WHITE/ORANGE

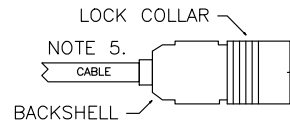


		FLOW SENSOR AND TRUE FLOW SENSOR MOUNTING DIAGRAM (BRACKET TYPE 2)			
		Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6			
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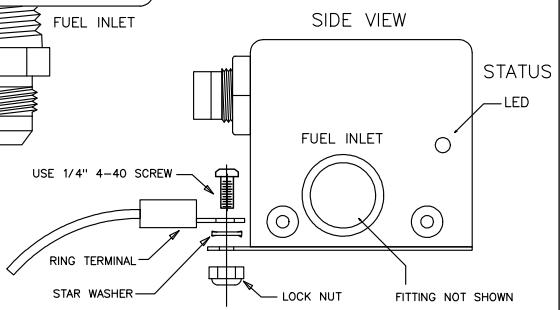
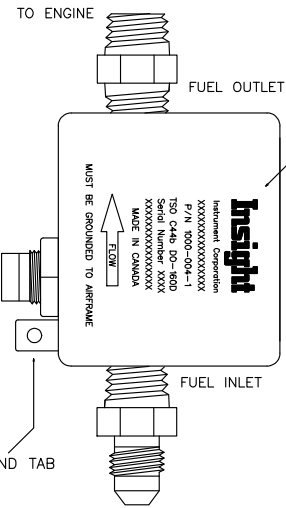
SEE PAGE ONE FOR REVISIONS.



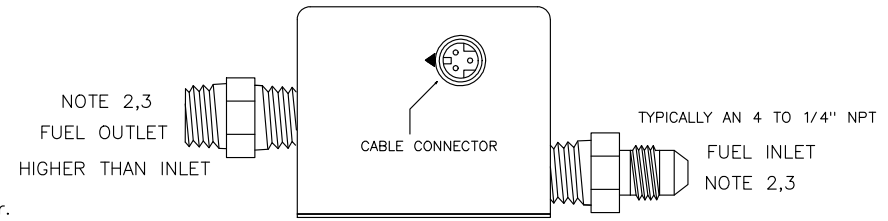
PUSH BACKSHELL TO CONNECT
PULL LOCK COLLAR TO DISCONNECT



SENSOR HOUSING
MUST BE GROUNDED TO AIRFRAME



FRONT VIEW

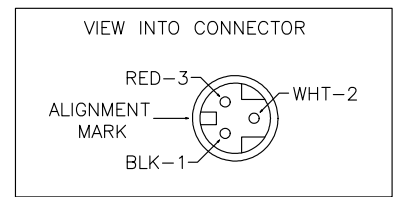


PART NUMBERS			
	INJECTOR	CARBURETOR	
FLOW SENSOR	1000-004-3(-Knnnnn)(-EPR)	1000-004-6(-Knnnnn)(-EPR)	
TRUE FLOW SENSOR	1000-006-3X SEE NOTE 12.	1000-006-6X	SEE NOTE 12.

NOTES:

1. Install INSIGHT FLOW SENSOR, one per engine, TWO per engine for pressure carburetor.
2. Use ONLY approved 1/4 NPT fittings for transducer inlet and outlet, installer supplied.
3. Use ONLY approved fuel line assemblies.
4. Use approved thread sealant. Torque pipe threads to a max of 15 ft. lbs. (180 in. lbs.)
5. INSIGHT supplied electrical harness with 3-pin connector, P/N 1000-809.
6. A straight length of fuel line upstream from the sensor inlet will improve accuracy by reducing turbulence in the rotor.
7. CONFIRM TRANSDUCER CASE IS GROUNDED TO AIRFRAME GROUND BEFORE POWERING UP.
8. Mount transducer with NAMEPLATE facing up.
9. Confirm all wiring PRIOR to powering up.
10. SEE INSTALLATION SCHEMATIC 1000-301 FOR FLOW SENSOR WIRING DETAILS.
11. SEE INSTALLATION SCHEMATIC 1000-816 FOR TRUE FLOW SENSOR WIRING DETAILS.
12. SPECIFY GPS MANUFACTURE AND MODEL WHEN ORDERING TRUE FLOW SENSOR, BENDIX KING B-FORMAT, APOLLO AND GARMIN Z-FORMAT.
13. FLOW SENSOR 30,000 PULSES PER GALLON UNLESS INDICATED BY -Knnnnn.
14. FLOW SENSOR DATA SIGNAL 5V UNLESS -EPR, EXTERNAL PULL-UP RESISTOR REQUIRED, SEE WIRING DIAGRAM.
15. Transducer self supporting on fuel fittings.

PIN	FUNCTION	WIRE COLOUR
1	COMMON (DO NOT GROUND)	BLACK OR WHITE/BLUE
2	SIGNAL OUT	WHITE
3	10 TO 30 VDC INPUT	RED OR WHITE/ORANGE



	FLOW SENSOR AND TRUE FLOW SENSOR MOUNTING DIAGRAM (BRACKET TYPE 3)			
	Insight Instrument Corporation Box 122, Fort Erie, Ontario L2A 5M6			
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		fmount3E	1000-303	[Signature] [Signature] [Signature] E
	SCALE	DATE	SHEET	REV
	none	June 1/99	3 OF 3	

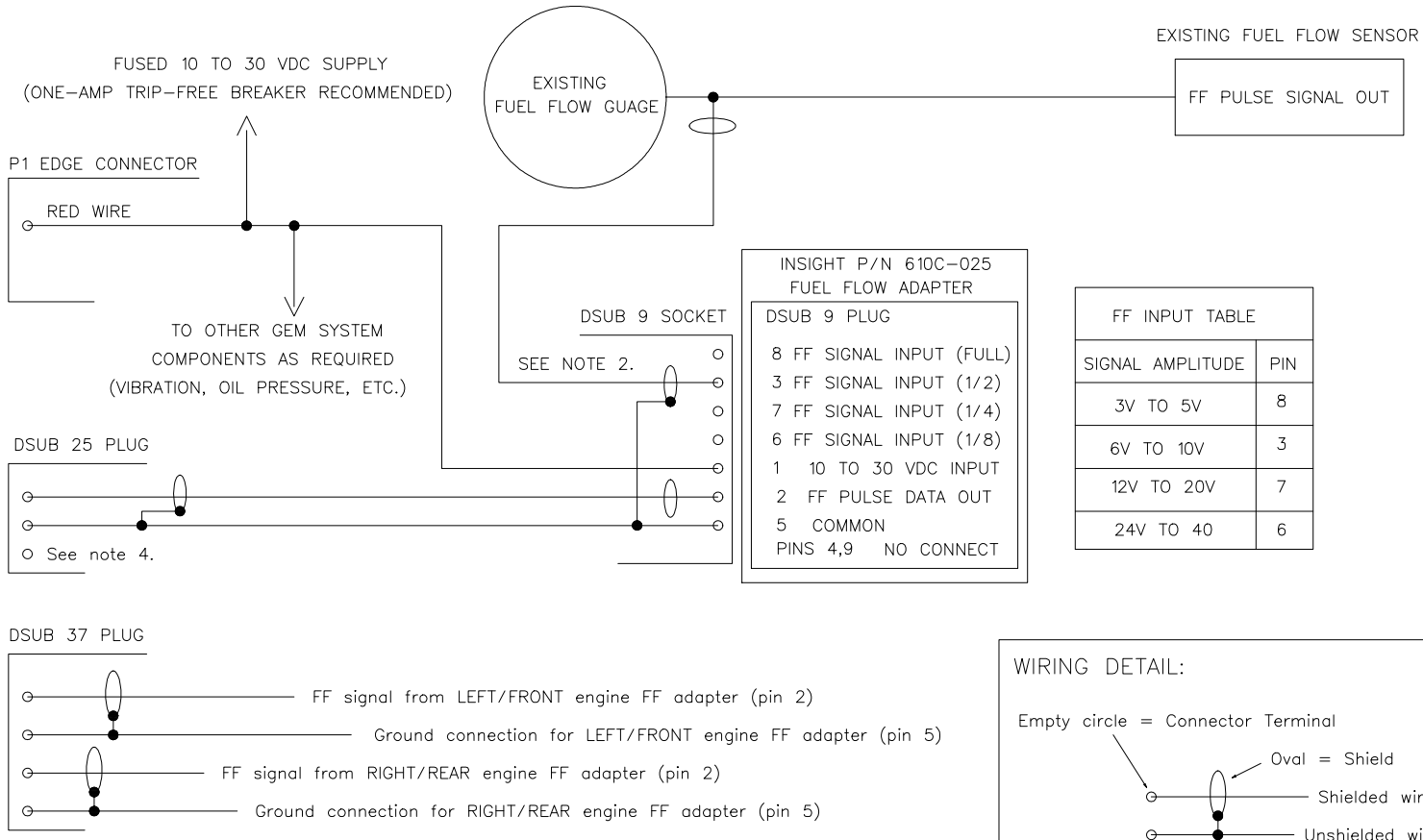
REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
None	First issue.	16 NOV 2010	JY
A	Changed connector from 15 to 9 pin.	15 DEC 2010	JY
C	Added detail for G4-Twin wiring.	2 MAR 2015	JY

P/N 610C-001 or 1200C-001
GRAPHIC ENGINE MONITOR

P1 (30 PIN EDGE)
All models.
10-30VDC INPUT PIN S

P2 (DSUB 25 SOCKET)
Pin numbers shown for
Single engine instruments.
Models: G2, G3, G4-Single.
FF1 23
DGND 25
FF2 12

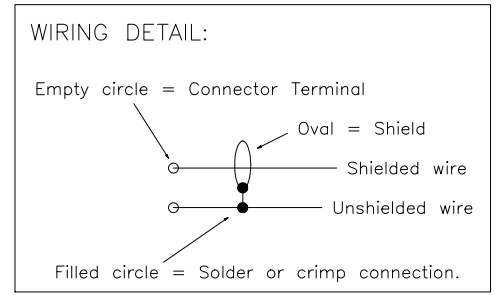
P2 (DSUB 37 SOCKET)
Pin numbers shown for
G4-Twin engine instruments.
FF1 (LEFT/FRONT) 11
DGND 30
FF2 (RIGHT/REAR) 16
DGND 17



INSIGHT P/N 610C-025
FUEL FLOW ADAPTER
DSUB 9 PLUG

8	FF SIGNAL INPUT (FULL)
3	FF SIGNAL INPUT (1/2)
7	FF SIGNAL INPUT (1/4)
6	FF SIGNAL INPUT (1/8)
1	10 TO 30 VDC INPUT
2	FF PULSE DATA OUT
5	COMMON
PINS 4,9 NO CONNECT	

FF INPUT TABLE	
SIGNAL AMPLITUDE	PIN
3V TO 5V	8
6V TO 10V	3
12V TO 20V	7
24V TO 40	6



NOTES:

1. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
2. Connect FF signal to one input pin, see input table.
3. Do not mount adapter in engine compartment.
4. FF2 for aircraft utilizing a fuel return line sensor, wire like FF1.
5. Twin engine aircraft require two adapters, one per engine.
6. For twin engine aircraft utilizing two single-engine GEM's (25 pin DSUB), wire each GEM/FF adapter as shown for single engine instruments.

<h1>Insight</h1>	FUEL FLOW ADAPTER WIRING DIAGRAM			
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2010	SIZE	FILE NAME: ffadaptC	DRAWING NO. 101116	DRN JY
	SCALE NONE	DATE 2 MAR 2015	SHEET 1 OF 1	REV C

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
None	Re-issue with new drawing number.	2 MAR 2015	JY

P/N 610C-001 or 1200C-001
GRAPHIC ENGINE MONITOR

P1 (30 PIN EDGE)
All models.
10-30VDC INPUT PIN S

P2 (DSUB 25 SOCKET)
Pin numbers shown for
Single engine instruments.
Models: G2, G3, G4-Single.

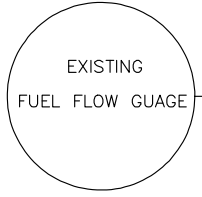
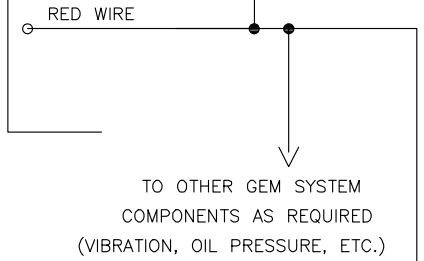
FF1	23
DGND	25
FF2	12

P2 (DSUB 37 SOCKET)
Pin numbers shown for
G4-Twin engine instruments.

FF1 (LEFT/FRONT)	11
DGND	30
FF2 (RIGHT/REAR)	16
DGND	17

FUSED 10 TO 30 VDC SUPPLY
(ONE-AMP TRIP-FREE BREAKER RECOMMENDED)

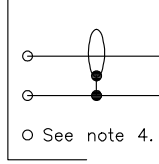
P1 EDGE CONNECTOR



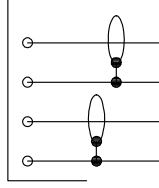
EXISTING FUEL FLOW SENSOR



DSUB 25 PLUG



DSUB 37 PLUG



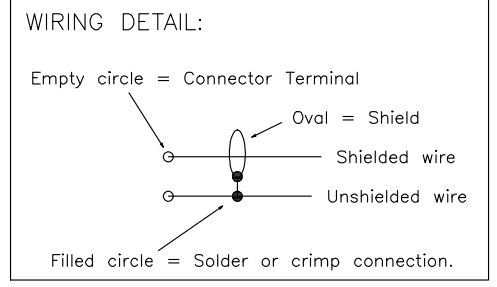
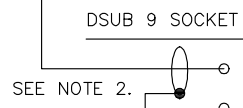
FF signal from LEFT/FRONT engine FF adapter (pin 6)
Ground connection for LEFT/FRONT engine FF adapter (pin 5)
FF signal from RIGHT/REAR engine FF adapter (pin 6)
Ground connection for RIGHT/REAR engine FF adapter (pin 5)

INSIGHT P/N 610C-025A
SENSITIVE FUEL FLOW ADAPTER

DSUB 9 PLUG

3	FF SIGNAL INPUT
8	FF SIGNAL INPUT
1	10 TO 30 VDC INPUT
6	FF PULSE DATA OUT
5	COMMON

PINS 2,4,7,9 NO CONNECT



NOTES:

1. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
2. Connect existing FF signal to PIN 3 OR 8 on FF adapter (interchangeable).
3. Do not mount adapter in engine compartment.
4. FF2 for aircraft utilizing a fuel return line sensor, wire like FF1.
5. Twin engine aircraft require two adapters, one per engine.
6. For twin engine aircraft utilizing two single-engine GEM's (25 pin DSUB), wire each GEM/FF adapter as shown for single engine installation.
7. For twin engine aircraft with a G4-Twin GEM (37 pin DSUB), connect the adapters as shown to the G4-Twin's 37 pin DSUB.

Insight

SENSITIVE FUEL FLOW ADAPTER
WIRING DIAGRAM

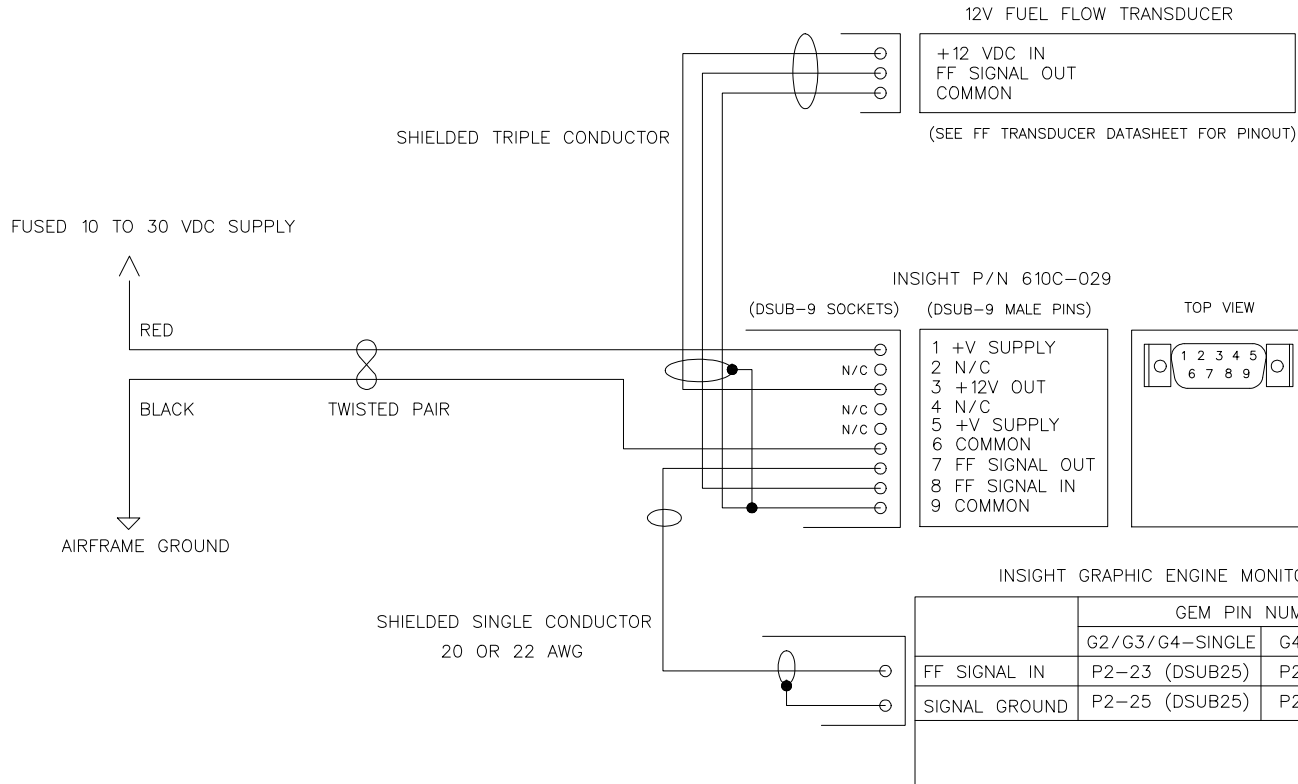
Insight Avionics Inc.
Box 122, Fort Erie, Ontario L2A 5M6

SIZE	FILE NAME:	DRAWING NO.	DRN	CKD	QCD	REV
NONE	SFFSAWD	150302	JY	JY	JY	-
SCALE	DATE	SHEET				
NONE	2 MAR 2015	1 OF 1				

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USE THIS DRAWING TO INSTALL A 12V FUEL FLOW TRANSDUCER IN A 28V BUSS AIRCRAFT USING AN INSIGHT P/N 610C-029 12V FUEL FLOW ADAPTER

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED



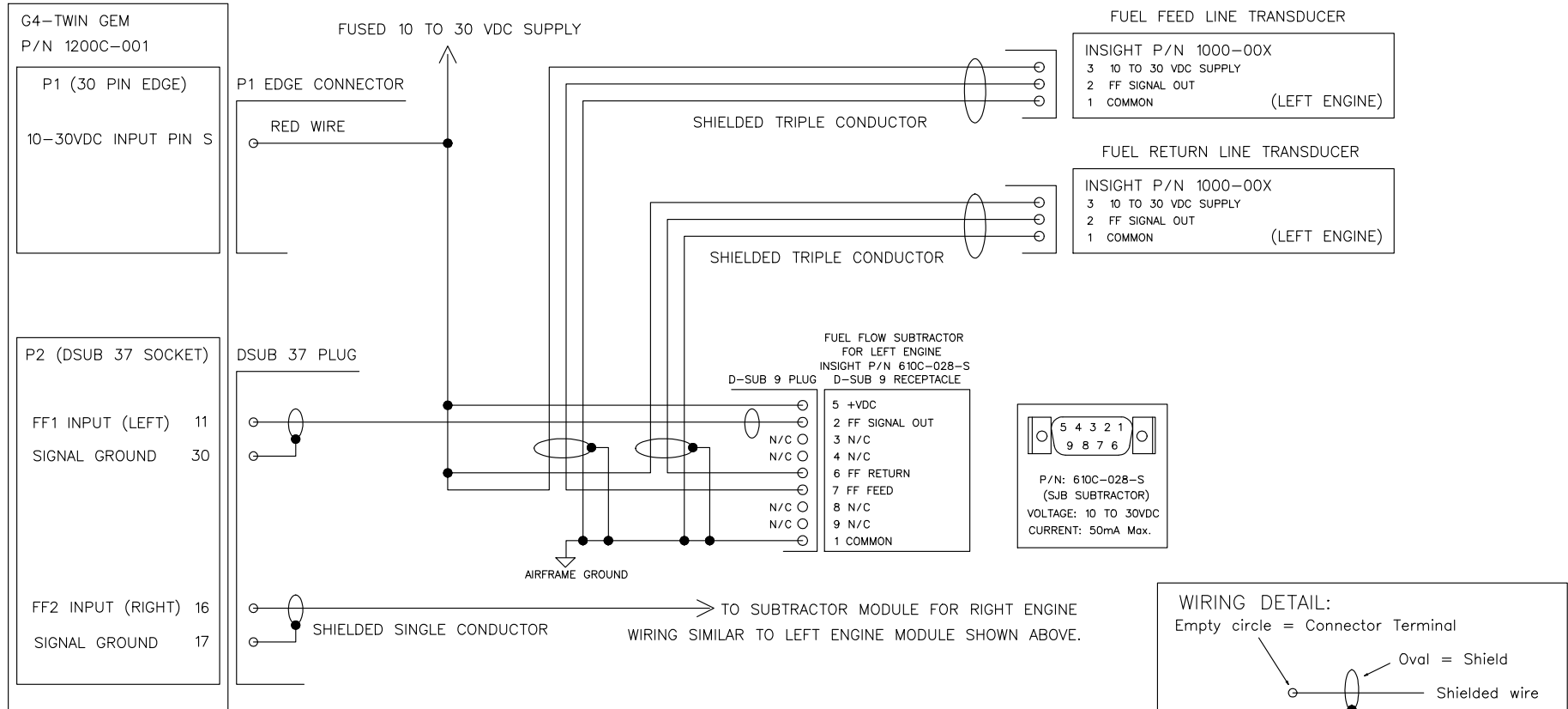
NOTES:

1. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
2. Module P/N 610C-029 has an internal pull-up resistor from signal (Pins 7,8) to +12V (Pins 1,5).
3. Do not mount 610C-029 module in engine compartment.
4. Use pigtail connections where multiple wires connect to a single pin.
5. Consult with Insight for equipment compatibility.
6. Connect the Red +V supply wire to the same +V supply as the GEM.

<h1>Insight</h1>	12V FUEL FLOW ADAPTER WIRING DIAGRAM			
	Insight Instrument Corp. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2013	SIZE A	FILE NAME: 12VFFWD	DRAWING NO. 130410	DRN <i>[Signature]</i>
SCALE NONE	DATE 10 APR 2013	SHEET 1	REV -	OCD <i>[Signature]</i>

FUEL FLOW SJB SUBTRACTOR WIRING DIAGRAM
 FOR TWIN-ENGINE A/C WITH PRESSURE CARBURETORS
 AND A G4-TWIN GRAPHIC ENGINE MONITOR P/N 1200C-001
 USE ONE SJB SUBTRACTOR PER ENGINE.

REVISIONS			
REV	DESCRIPTION	DATE	APPROVED



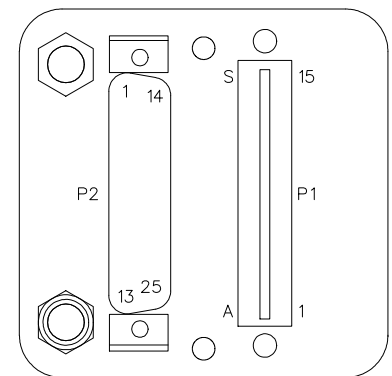
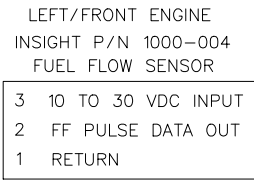
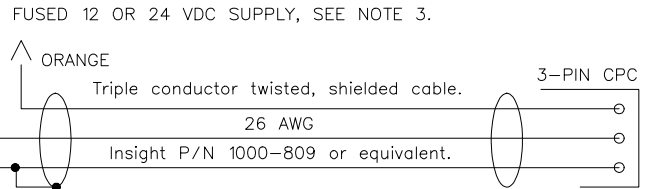
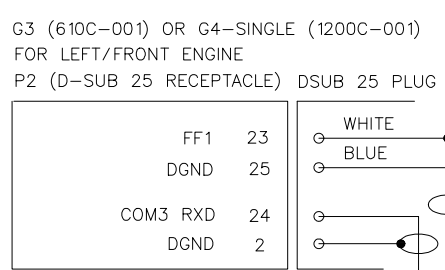
NOTES:

1. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
2. Consult with Insight for equipment compatibility.
3. Do not mount Subtractor modules in engine compartment(s).
4. Use pigtail connections where multiple wires connect to a single pin.
5. Compatible fuel flow sensors may be substituted.

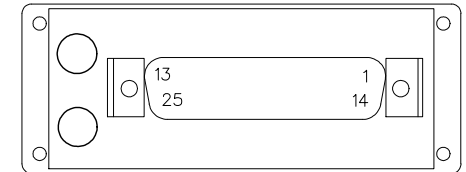
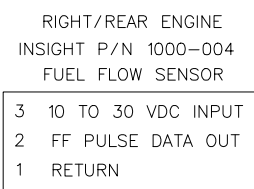
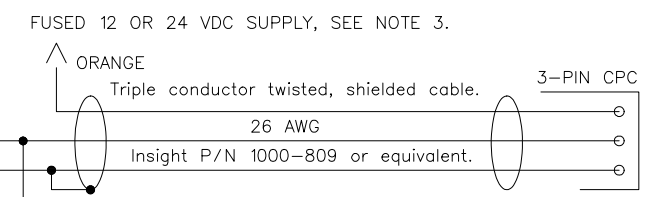
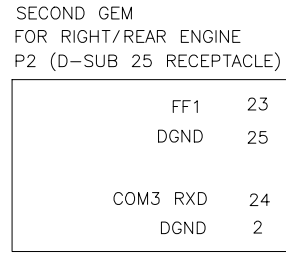
		FUEL FLOW SUBTRACTOR WIRING DIAGRAM			
		Insight Instrument Corp. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2013	SIZE: A SCALE: NONE	FILE NAME: ffsubwd DATE: 10 APR 2013	DRAWING NO.: 130523 SHEET: 1 OF 1	DRN: [Signature] CKD: [Signature] QCD: [Signature] REV: -	

FUEL FLOW AND AIRDATA INTERFACE (TAS-1000)
 FOR SINGLE ENGINE G-SERIES GEMS
 (FOR G4-TWIN GEM INTERFACE SEE DRAWING NUMBER G4-315)

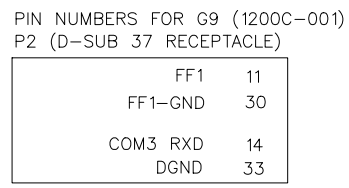
REVISIONS			
REV	DESCRIPTION	DATE	APPROVED
A	See ECNs for changes.	10 DEC 12	JY
B	Added pin numbers for G9.	1 OCT 13	JY



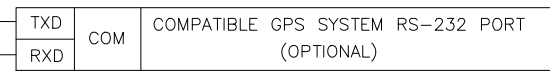
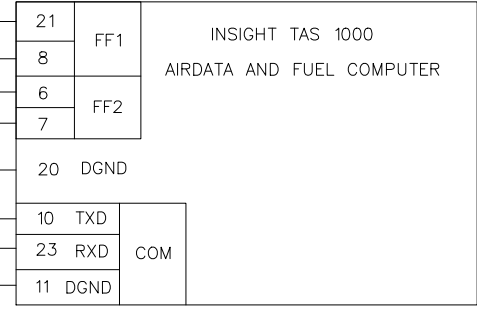
G3 (610C-001) REAR VIEW
 G4-SINGLE (1200C-001) SIMILAR



TAS-1000 REAR VIEW



WIRING SIMILAR TO ABOVE



NOTES:

1. This drawing is for single or twin-engine aircraft, twin configuration shown.
2. For single engine aircraft omit second GEM and FF sensor.
3. Power FF sensor from same supply as Graphic Engine Monitor.
4. Use 24 AWG copper wire with TEFZEL or equivalent insulation unless otherwise specified.
5. Use pigtail connections where multiple wires connect to a single pin.
6. See TAS-1000 Installation Manual for additional TAS and GPS data.

	GEM/FF/TAS AND GPS INTERCONNECTION			
	Insight Avionics Inc. Box 122, Fort Erie, Ontario L2A 5M6			
This drawing is the property of Insight Instrument Corporation and contains confidential and proprietary information. Reproduction or unauthorized use is strictly forbidden. © Copyright 2012	SIZE: 2G3&TAS SCALE: NONE	FILE NAME: 2G3&TAS DATE: 1 OCT 2013	DRAWING NO.: 12 12 10 SHEET: 1 OF 1	DRN: [Signature] CKD: [Signature] QCD: [Signature] REV: B

Technical Support

If you have difficulty installing or using a G Series system, please read the G series documentation. Every G Series system is shipped with complete instructions for installation and use on a DVD. You may also find the same information on our website www.insightavionics.com under Documentation.

The answers to many technical questions can be found in these documents. Insight provides customer support free of charge for as long as you own the instrument. If you have any questions concerning G Series operation do not hesitate to call.

The Customer Service department accepts calls Monday through Friday between 9 am and 5 pm EST. Be sure to have your instrument model number and serial number(s) ready when you call.

GEM Model No. _____

GEM Serial No. _____

Aircraft Type _____

Service Procedures

Like many modern electronic devices, the Graphic Engine Monitor is extremely reliable. Other than initial configuration during installation, the G series requires no adjustment or routine maintenance. The G Series routinely performs a self-test on power-up to test its vital functions. Should the G series fail this test, it will halt on a red screen with a description of the error.

Errors of this nature must be returned to the factory for service.

If the instrument performs the self-test successfully but you feel that the instrument is not functioning as expected discuss the problem with your dealer, or consult the troubleshooting section of the G Series Installation Instructions. Keep in mind that in the vast majority of cases, erratic or unusual G Series operation can be traced to an installation problem, a problem with probes, wiring harness or the aircraft's electrical system.

Due to its reliability and the extensive testing that it undergoes, the instrument itself is the least likely source of trouble. For this reason we strongly encourage you to contact Insight Customer Service at one of the numbers listed below before returning an instrument to the factory, or any time you have any questions concerning the operation of the Graphic Engine Monitor. By contacting us first we may be able to assist you in correcting your problem without the hassle and downtime of sending in an instrument that is not the source of the difficulty.

(905) 871-0733

Web: www.insightavionics.com

E-mail: techsupport@insightavionics.com